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NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TECHNICAL NOTE

No. 1143

CHARTS FOR DETERMINING THE CHARACTERISTICS OF SHARP-NOSE AIRFOILS IN TWO-DIMENSIONAL FLOW AT SUPERSONIC SPEEDS

By H. Reese Ivey, George W. Stickle,
and Alberta Schuettler

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Langley Field, Va.

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SUMMARY

Solutions of the Hugoniot shock equations and Meyer expansion equations are plotted in such a manner as to permit the pressure distribution, the local Mach number, and the angles of shock waves on arbitrary sharp-nose airfoils at supersonic speeds to be obtained directly.

INTRODUCTION

Ackeret, in reference 1, gives a method for calculating the pressure distribution over thin, sharp, two-dimensional airfoils at supersonic speeds. This method, based on the theory of small disturbances, is only a first approximation and therefore is most accurate for thin airfoils.

The exact relationship for the pressure rise through a normal shock wave, as given by Hugoniot, is discussed in reference 2. According to reference 3, the corresponding relations which apply directly to the pressures on a straight surface of an airfoil immediately behind an oblique shock were obtained by Meyer as early as 1908. A discussion of Meyer's equations for the expansion of supersonic flow around an infinite corner is also given in reference 3. Frequently, interference exists between

shock and expansion waves caused by the intersection of two or more of these waves. When this intersection is close to the airfoil, as, for instance, when the airfoil has considerable curvature, the calculations yielded by the aforementioned equations are not exact.

It has been shown by Ferri (reference 4) that the equations for an oblique shock combined with the expansion equations give a close approximation to experimental results as reviewed in the section "Presentation of Figures" in this report. The use of the equations, however, involves long and difficult computations. The purpose of this paper is to give graphic solutions of these equations in a form suitable for rapid calculation. Because the size of the graphs limits their accuracy, tables are given from which computational graphs of much greater accuracy may be plotted. The relations given herein apply directly to a two-dimensional, or cylindrical, flow in which the transverse velocity is supersonic. As pointed out by Buserann (reference 5) they may be adapted to the case of oblique motion of the cylindrical airfoil by the addition of an arbitrary axial velocity. Thus, as in reference 5, the relations may be applied to the case of a swept-back airfoil lying ahead of the Mach lines, in which case the velocities and Mach numbers used in the calculation are those corresponding to the transverse component of the flight velocity. In case the airfoil is swept behind the Mach lines the flow will be of a different type as discussed in reference 6.

SYMBOLS

M	Mach number
p	static pressure
q	dynamic pressure
β	change in direction of flow (see fig. 1)
γ	ratio of specific heat at constant pressure to specific heat at constant volume = 1.4 for air
θ	angle of shock wave relative to direction of flow before shock

v angle around which the flow would have to expand from $M = 1$ to the given local Mach number

p density of gas

Subscripts:

a for conditions after a disturbance

b for conditions before a disturbance

n for local condition under consideration

o free stream

PRESENTATION OF FIGURES

A supersonic two-dimensional air flow around an airfoil may change its direction either by deflection or by expansion around a corner. In case the change in air-flow direction occurs by deflection, a shock wave is set up, and in case the change is by expansion, an expansion wave is set up. In either case, the change of state of the gas can be presented as a function of the local Mach number before the disturbance and the change in direction of the gas.

The equations from which the charts presented herein are derived are given in the appendix. Values of local Mach number, pressure ratio, and pressure coefficient across shock waves are given in table I. The local Mach numbers before and after expansions are presented in table II and the static relations across expansion waves, in table III. Table IV gives the pressure ratios based on free-stream dynamic pressure for various Mach numbers.

Before the method of determining pressure distribution, lift, drag, and moments may be discussed, a method of measuring the angles that cause expansions and shocks must be selected. Figure 1 shows the method used in the present paper for measuring angles causing expansions; figure 2 shows that for measuring angles causing shocks. The angle causing the disturbance is designated β in both cases; β is considered negative if the disturbance set up is an expansion wave and positive if it is a shock wave.

Figure 3 shows the manner in which the flow changes when the angles are made too large for the given speeds. If the angle causing the shock is too great, the shock wave separates from the airfoil surface. In figure 4 are given the maximum angles that may exist before the shock wave separates, calculated as the boundary condition between the region giving two solutions and the region for which no solution exists. If the trailing portion of the airfoil is too blunt, the flow may separate from the airfoil and leave a turbulent wake. Figure 3(b) shows that the expansion of the flow outside the wake is actually less than it would have been had it followed the surface. The pressure on the back of the airfoil does not decrease so much as it would if the flow failed to separate. The drags calculated if no separation is assumed will therefore be higher than those actually experienced.

The local Mach number after a disturbance (shock or expansion) is shown in figure 5 to be a function of the local Mach number before the disturbance and the angle causing the disturbance. For example, if a flow at a Mach number of 4.0 impinges on a surface set at 5° to that flow, a shock wave is set up behind which the local Mach number is 3.64, while the flow behind the shock wave is parallel to the surface. On the other hand, the same flow expanding around a 5° corner produces a local Mach number equal to 4.4 on the surface behind the expansion.

Figure 6 gives the ratio of static pressures across shock and expansion waves. For example, assume that a flow at a local Mach number of 4.0 is shocked by a surface slope change of $\beta = 5^\circ$. From figure 6 the pressure ratio across the shock is 1.61, which means that the pressure is much higher on the surface behind the shock than on the surface before it. If the flow at $M = 4.0$ had expanded 5° , then the pressure would have dropped to 0.588. From this example it is seen that the 5° shock increased the pressure by 61 percent, whereas the 5° expansion decreased the pressure only 41 percent. Ackeret's method in reference 1 predicts equal changes in pressure for both the shock and the expansion, since it is based on small disturbances. Present results indicate, therefore, that angles as large as 5° require a more accurate approximation than that given by Ackeret.

The use of figures 5 and 6 can be demonstrated by solving for the local Mach numbers and pressures on the simple airfoil shown in figure 7. The coordinates of figures 5 and 6 are based on conditions before the disturbance and conditions after the disturbance. In figure 7 the conditions after one disturbance are noted to be the conditions before another disturbance. The numerical subscripts found in the symbols of figure 7 are to be associated, therefore, for use in the charts of figures 5 and 6, with the subscripts a and b, according to their relative positions with respect to the disturbance. The airfoil of figure 7 is a symmetrical, double-wedge airfoil having a 2° included angle at the leading and trailing edges. For use in this example the airfoil is at a positive angle of attack of 3° and is moving at a free-stream Mach number of 4.0. The pertinent angles as well as the conditions to be determined are shown on figure 7.

Enter figure 5 at $M_0 = 4.0$ and $\beta_1 = -2^{\circ}$, and read off $M_1 = 4.16$. This Mach number is used to obtain $M_2 = 4.33$. Values for the lower surface of the airfoil are obtained in a similar manner. At coordinates of $M_0 = 4.0$ and $\beta_3 = 4^{\circ}$, $M_3 = 3.70$; and, similarly, when the flow at M_3 is expanded 2° , M_4 is found to be 3.84. A shock wave and an expansion wave are shown at the trailing edge; however, since these disturbances do not affect the pressures on the airfoil, they will be neglected, and M_1 , M_2 , M_3 , and M_4 are the only Mach numbers which are discussed.

The pressure ratios across the shock and expansion waves can be determined from figure 6. Enter figure 6 at $M_0 = 4.0$ and an expansion angle of 2° and read

$\frac{p_1}{p_0} = 0.817$. At coordinates of $M_1 = 4.16$ and $\beta_2 = -2^{\circ}$,

$\frac{p_2}{p_1}$ is found to be 0.809. Then

$$\frac{p_2}{p_0} = \frac{p_1}{p_0} \frac{p_2}{p_1} = 0.817 \times 0.809 = 0.661$$

For the lower surface of the airfoil, p_3/p_0 is found at coordinates $M_0 = 4.0$, $\beta_3 = 4^\circ$ to be $\frac{p_3}{p_0} = 1.47$. At $M_3 = 3.70$, $\beta_4 = -2^\circ$, $\frac{p_4}{p_3} = 0.829$. Then

$$\frac{p_4}{p_0} = \frac{p_4}{p_3} \times \frac{p_3}{p_0} = 1.219$$

The pressure ratios $\frac{p_a}{p_b}$ may be converted to local pressure coefficients $\frac{\Delta p}{q_\infty}$ by the use of the plot given in figure 8. The results obtained for the local Mach numbers, pressure ratios, and pressure coefficients are illustrated in figure 9.

Once the pressure distribution is determined, the lift, drag, and moment coefficients can be obtained by integrating plots of the types given in figures 10, 11, and 12. The lift coefficient is obtained by integrating the projection of the airfoil pressure distribution on a plane parallel to free-stream direction. For the example airfoil at Mach number 4.0 and angle of attack 3° , the lift coefficient is 0.0540. The drag coefficient is found in the same manner except that the integration is over the projection of the airfoil pressure distribution on a plane perpendicular to free stream. The pressure drag coefficient for the example airfoil is 0.00315. The total section drag coefficient is the sum of the viscous and pressure-drag coefficients; for instance, if the viscous-drag coefficient is 0.0060, then the total drag coefficient is $0.0060 + 0.00315 = 0.00915$. The moment coefficient, obtained by integrating the elemental moments about the point desired, becomes 0.001112 when taken about the center for the example airfoil.

In the preceding examples step-by-step calculations were made along the airfoil, in which case the results obtained at any point are dependent on the accuracy of those at the preceding points. The results thus obtained on the rear of the airfoil may be subject to greater inaccuracies than are necessary. A method is consequently

given for determining the conditions behind each of a series of expansions independently of the conditions existing at intermediate points. For the example airfoil of figure 7, the free-stream flow was expanded around a 2° corner to obtain the conditions on the front of the upper surface and these conditions were then expanded around the second 2° corner to give the conditions on the rear of the upper surface. These last conditions, however, could have been found directly by referring the rear of the upper surface to the free-stream conditions and expanding through the total angle (4°) at once. Theoretically the results obtained are exactly the same regardless of which method is used, provided no shock waves are present between the end-points of the calculation. This method of adding angles does not apply when there are intermediate shock waves because of loss of total head in the shock wave.

If it is desired to calculate only the pressure distribution, it is not necessary to find M_1 , M_2 , $M_{1\perp}$, or p_2/p_1 for an airfoil similar to the type in the example given.

Figure 13 is taken from reference 4 to compare the experimental pressure distribution on an airfoil with the calculated distribution. Even though the wind-tunnel tests were of a very small model and although the airfoil is not of a type particularly suitable for calculations, the calculated and experimental values seem to compare favorably except for the region of separated flow near the upper trailing edge of the airfoil. The method of the present report is not exact for an airfoil of this type, which has considerable curvature along its entire length. The inaccuracy caused by the curvature, however, seems to be small. The thickness ratio and angle of attack of the example airfoil are somewhat higher than those for which the method is recommended. Reference 4 explains the separated region of flow.

Langley Memorial Aeronautical Laboratory
National Advisory Committee for Aeronautics
Langley Field, Va., April 4, 1946

APPENDIX

METHOD OF ANALYSIS

Shock Waves

Supersonic air flow about an airfoil may be said to consist of expansions and shocks. Reference 2 mentions the fact that a change in entropy occurs through a shock wave. Three conditions are shown, however, to apply to the velocities, pressures, and densities at the two sides of the shock wave, namely:

- (a) Continuity of mass
- (b) Balance between pressure difference and change of momentum
- (c) Conservation of energy

These conditions lead to the three basic equations:

or

$$\left. \begin{aligned} \frac{\rho_a}{\rho_b} &= \frac{\tan \theta}{\tan (\theta - \beta)} \\ \Delta P &= \frac{\sin \beta}{\sin \theta \times \cos (\theta - \beta)} \end{aligned} \right\} \quad (1)$$

$$\left. \begin{aligned} \frac{\Delta P}{P_b} &= 2 \sin^2 \theta \frac{\Delta P}{P_a} \\ &= \frac{2 \sin \beta \times \sin \theta}{\cos (\theta - \beta)} \end{aligned} \right\} \quad (2)$$

$$\frac{\Delta p}{\Delta \rho} = \gamma \frac{p_b + \frac{\Delta p}{2}}{p_b - \frac{\Delta p}{2}} \quad (3)$$

Then, by use of the relation

$$\frac{\Delta p}{q_b} = \frac{2}{\gamma M_b^2} \left(\frac{p_a}{p_b} - 1 \right) \quad (4)$$

it follows that

$$\frac{p_a}{p_b} = \frac{\gamma M_b^2 \sin \beta \sin \theta}{\cos(\theta - \beta)} + 1 \quad (5)$$

$$\frac{1}{M_b^2} = \sin^2 \theta - \frac{\gamma + 1}{2} \frac{\sin \beta \sin \theta}{\cos(\theta - \beta)} \quad (6)$$

$$M_a = M_b \frac{\cos \theta}{\cos(\theta - \beta)} \sqrt{\frac{p_b}{p_a} \frac{\rho_a}{\rho_b}} \quad (7)$$

By substitution of arbitrary values of θ and β in equations (1) and (6), the corresponding values of density ratio across the shock and Mach number before the shock are obtained. If the simultaneous values of θ , β , and M_b are used with equations (5) and (7), the pressure ratio across the shock and the Mach number after the shock are obtained.

Figure 14 shows the angle of the shock wave as a function of the Mach number before the shock M_b and the angle defining the change in direction of the flow β .

The pressure ratios and the Mach numbers after the shock have already been discussed for figures 5 and 6. Use of the ratio of pressure after any shock wave to free-stream static pressure, together with free-stream Mach number in equation (4), makes possible the determination of the pressure coefficient behind that shock wave. Figure 8 has shown the graph for converting pressure ratios to pressure coefficients.

Expansion Waves

The flow after the shock wave may be considered adiabatic as long as the flow is expanding. By the use of such flow conditions, the velocities, densities, and pressures may be calculated. Experimentally some trouble is encountered when extremely large angles of expansion are used. The flow may break down and form a turbulent wake of somewhat higher static pressure than might be expected if the flow had continued to expand around the corner.

Reference 3 considers that a flow at a Mach number of 1 expands around some angle v and reaches a higher Mach number M defined by the relation

$$v = \sqrt{\frac{\gamma + 1}{\gamma - 1}} \tan^{-1} \left(\sqrt{M^2 - 1} \sqrt{\frac{\gamma - 1}{\gamma + 1}} \right) - \cos^{-1} \frac{1}{M} \quad (8)$$

By expanding around an angle v_b , the flow reaches a Mach number M_b ; and by expanding around some larger angle v_a , the flow reaches some higher Mach number M_a . A flow at the first Mach number M_b can then reach the higher Mach number M_a by expanding around the small angle,

$$-\beta = v_a - v_b \quad (9)$$

Equations (8) and (9) serve as the basis for calculating the expansion lines in figure 5 showing the variation of local Mach number with change in surface slope.

Another equation derived from the work of reference 3 gives the pressure ratio across expansion waves as

$$\frac{p_a}{p_b} = \left[\frac{2 + (\gamma - 1) M_b^2}{2 + (\gamma - 1) M_a^2} \right]^{\frac{\gamma}{\gamma-1}} \quad (10)$$

From equations (5), (9), and (10), it is possible to calculate the part of figure 6 that gives the pressure ratio across expansion waves as a function of the local Mach number before the expansion and the change in surface slopes.

The figures shown in this report, because of their limited size, may not be accurate enough for routine calculations. It may be desirable to plot the graphs to a larger scale before using them. For this reason the values are listed in tabular form for the main graphs.

Tables I and II should be accurate to all the figures shown, but table II may not be exact in the last figure since the expansion calculations required graphical interpolation between very close computed points.

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use supplementTABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO,
AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES

θ (deg)	β (deg)	M_b	M_a	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$	θ (deg)	β (deg)	M_b	M_a	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_a}$
8 ↓	0	7.18546	7.18546	1.00000	0	16 ↓	6	4.94653	4.35137	2.00221	0.05851
	1	7.80125	7.54749	1.20554	.00490		7	5.32290	4.55154	2.35426	.06805
9 ↓	2	8.60430	8.07261	1.50622	.00977	17 ↓	8	5.82325	4.79190	2.83915	.07718
	3	6.39261	6.39261	1.00000	0		9	5.47624	5.08655	3.55104	.08689
10 ↓	0	6.67407	6.69357	1.18235	.00551	18 ↓	10	7.40814	5.45674	4.69797	.09626
	1	7.16067	7.06779	1.45092	.01100		11	5.90088	5.95848	6.85594	.10559
11 ↓	2	8.27634	7.54385	1.78981	.01646	19 ↓	1	3.42032	3.42032	1.00000	0
	3	5.75876	5.75876	1.00000	0		2	3.55532	3.19188	1.09391	.01061
12 ↓	4	6.11659	6.29875	1.16236	.00614	20 ↓	3	5.70612	5.77318	1.20313	.02113
	5	6.62119	6.28816	1.37585	.01224		4	3.87615	3.66580	1.33170	.03124
13 ↓	6	7.22344	6.64849	1.66896	.01832	21 ↓	5	4.07015	3.77171	1.18543	.04186
	7	8.02263	7.10485	2.09762	.02136		6	4.29155	3.89345	1.67261	.05210
14 ↓	8	5.24082	5.24082	1.00000	0	22 ↓	7	4.55860	4.03450	1.90576	.06227
	9	5.55967	5.15348	1.14629	.00676		8	4.87573	4.19924	2.20412	.07236
15 ↓	10	5.91262	5.66635	1.33358	.01369	23 ↓	9	5.26736	4.39418	2.60026	.08210
	11	6.12373	5.95318	1.58610	.02029		10	5.76803	4.62799	3.15127	.09237
16 ↓	12	7.01599	6.29392	1.92121	.02682	24 ↓	11	6.44180	4.91440	3.97167	.10230
	13	7.83581	6.73948	2.44137	.03354		12	7.41936	5.27598	5.32300	.11219
17 ↓	14	8.06358	7.30617	3.26355	.04004	25 ↓	13	9.05211	5.73979	7.06900	.12204
	15	9.80977	4.80977	1.00000	0		14	3.23604	3.23604	1.00000	0
18 ↓	16	5.07739	4.96951	1.13334	.00739	26 ↓	1	3.57720	3.29834	1.08900	.01128
	17	5.39281	5.15796	1.29999	.0174		2	3.49146	3.36878	1.19115	.02214
19 ↓	18	5.77302	5.38268	1.51111	.02204	27 ↓	3	3.64174	3.44868	1.31087	.03349
	19	6.24375	5.65433	1.79936	.02929		4	3.81150	3.59348	1.45184	.04443
20 ↓	21	6.81911	5.98883	2.19908	.03652	28 ↓	5	4.00564	3.61318	1.62091	.05528
	22	6.66903	7.11123	2.73939	.0471		6	4.23683	3.76221	1.82753	.06604
23 ↓	23	8.07172	6.96292	3.80282	.05087	29 ↓	7	4.57062	3.84945	2.16071	.07937
	24	6.81911	5.98883	2.19908	.03652		8	4.81743	3.05749	2.11888	.08784
25 ↓	25	7.66903	7.11123	2.73939	.0471	30 ↓	9	5.21582	4.25058	2.86418	.09789
	26	8.07172	6.96292	3.80282	.05087		10	5.72912	4.47828	3.49010	.10838
27 ↓	27	6.10349	5.39892	2.03258	.03660	31 ↓	11	6.42781	4.75694	4.13641	.11882
	28	6.71567	5.72228	2.49588	.04738		12	7.46204	5.10668	6.03381	.12921
29 ↓	29	7.55459	6.12986	3.20247	.05555	32 ↓	13	9.22255	5.56127	9.30925	.13956
	30	8.80760	6.66172	4.41302	.06285		14	3.07154	3.07154	1.00000	0
31 ↓	31	4.14543	4.14543	1.00000	0	32 ↓	1	3.18101	3.12594	1.08469	.01195
	32	4.67339	4.57891	1.12271	.00803		2	3.30163	3.18741	1.18134	.02377
33 ↓	33	4.97854	4.73465	1.27307	.01600	33 ↓	3	3.43552	3.25672	1.29288	.03515
	34	5.24983	4.91619	1.46158	.02392		4	3.58559	3.35514	1.42320	.04702
35 ↓	35	5.63112	5.13596	1.70535	.03178	34 ↓	5	3.75551	3.42421	1.57743	.05849
	36	6.10349	5.39892	2.03258	.03660		6	3.95028	3.52573	1.76303	.06985
37 ↓	37	6.71567	5.72228	2.49588	.04738	35 ↓	7	4.17663	3.61210	1.99073	.08115
	38	7.55459	6.12986	3.20247	.05555		8	4.44535	3.77664	2.27703	.09232
39 ↓	39	8.80760	6.66172	4.41302	.06285	36 ↓	9	4.77075	3.93551	2.64786	.09450
	40	4.13360	4.13360	1.00000	0		10	5.17739	4.11880	3.14812	.11448
41 ↓	41	4.33044	4.24218	1.11376	.00567	37 ↓	11	5.70582	4.30478	3.85929	.12547
	42	4.55638	4.37682	1.25087	.01726		12	6.43041	4.61223	1.95273	.13610
43 ↓	43	4.81978	4.52876	1.41949	.02580	38 ↓	13	7.55249	4.95273	6.04971	.14729
	44	5.13240	4.70707	1.63192	.03427		14	9.48526	5.39507	10.95449	.15813
45 ↓	45	5.51261	4.91882	1.90826	.04270	39 ↓	15	3.92381	3.92381	1.00000	0
	46	5.98881	5.17368	2.28225	.05107		16	3.02354	3.57148	1.08078	.01263
47 ↓	47	6.62228	5.19587	2.81769	.05921	40 ↓	17	3.15239	3.62192	1.17239	.02510
	48	7.47150	5.88074	3.64800	.06771		18	3.25271	3.08575	1.27725	.03714
49 ↓	49	8.79126	6.39434	5.11042	.07598		19	3.38650	3.15592	1.39317	.04961
	50	9.06369	3.86369	1.00000	0	41 ↓	20	3.53576	3.25092	1.54042	.06172
51 ↓	51	4.05562	3.95393	1.06114	.00931		21	3.70716	3.31824	1.70890	.07369
	52	4.16555	3.88289	1.23231	.01531	42 ↓	22	3.90296	3.41765	1.91224	.08555
53 ↓	53	4.45565	4.19791	1.38489	.02770		23	4.02354	3.51597	2.16293	.09732
	54	4.71842	4.34601	1.57529	.03679	43 ↓	24	4.48526	5.39507	2.89260	.12061
55 ↓	55	5.03145	4.51957	1.81179	.04587		25	5.15114	3.99719	3.45455	.13215
	56	5.41100	5.72539	2.12408	.05479	44 ↓	26	5.69739	4.23372	4.28333	.14362
57 ↓	57	5.88618	5.04261	2.55029	.06370		27	6.16071	4.47810	5.52986	.15593
	58	6.55211	5.27618	3.16796	.07258	45 ↓	28	7.63854	4.81027	7.79624	.16640
59 ↓	59	7.12624	5.65781	4.14339	.08113		29	8.53090	5.24117	13.02305	.17772
	60	8.82181	6.14628	5.91549	.09023	46 ↓	30	4.73461	5.81626	2.89260	.12061
61 ↓	61	3.62792	3.62792	1.00000	0		31	5.15114	5.6507	2.17965	.10961
	62	3.77964	3.71061	1.09962	.00996	47 ↓	32	5.69739	4.23372	4.28333	.14362
63 ↓	63	3.95037	3.80507	1.21661	.01983		33	6.16071	4.47810	5.52986	.15593
	64	4.11479	3.91330	1.35610	.02961	48 ↓	34	7.63854	4.81027	7.79624	.16640
65 ↓	65	4.36900	4.03782	1.52531	.03931		35	8.53090	5.24117	13.02305	.17772

TABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO,
AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Continued

θ (deg)	β (deg)	k_b	M_a	$\frac{p_a}{p_b}$	$\frac{\Delta p_{a,b}}{q_0}$	θ (deg)	β (deg)	k_b	M_a	$\frac{p_a}{p_b}$	$\frac{\Delta p_{a,b}}{q_0}$
21	0	2.79041	2.79041	1.00000	0	25	0	2.36619	2.36619	1.00000	0
	1	2.88143	2.83236	1.07736	.01331		1	2.43311	2.39158	1.06692	.016149
	2	2.98066	2.87953	1.16150	.02645		2	2.50490	2.42023	1.11078	.032053
	3	3.08951	2.93252	1.26352	.03944		3	2.58225	2.45239	1.22268	.07707
	4	3.20981	2.99660	1.37704	.05228		4	2.66584	2.48812	1.31421	.063162
	5	3.34375	3.05910	1.50857	.06498		5	2.75692	2.52825	1.41711	.078398
	6	3.47911	3.12563	1.64697	.07636		6	2.85666	2.57296	1.53377	.093442
	7	3.66576	3.22034	1.81478	.09002		7	2.96666	2.62291	1.66728	.10831
	8	3.86318	3.31777	2.06948	.10237		8	3.08899	2.67881	1.82163	.12301
	9	4.09428	3.42921	2.34504	.11463		9	3.22162	2.74136	2.00218	.13755
	10	4.37040	3.55785	2.69520	.12679		10	3.38185	2.81164	2.21653	.15196
	11	4.70837	3.70460	3.15498	.13887		11	3.56060	2.89100	2.47509	.16622
	12	5.13650	3.88129	3.78653	.15088		12	3.76908	2.98103	2.79352	.18036
	13	5.70365	4.09575	4.70770	.16282		13	4.01684	3.08381	3.19547	.19438
	14	6.50821	4.22408	6.17984	.17470		14	4.31849	3.20211	3.71940	.20831
	15	7.78248	4.6778	8.90683	.18653		15	4.69712	3.33960	4.43072	.22214
22	0	2.66944	2.66944	1.00000	0	26	0	2.28118	2.28118	1.00000	0
	1	2.75318	2.70503	1.07435	.014012		1	2.34464	2.30339	1.06582	.01710
	2	2.84397	2.74795	1.15755	.027827		2	2.41071	2.32980	1.13624	.03319
	3	2.94308	2.79453	1.25145	.041472		3	2.48251	2.35720	1.21505	.04985
	4	3.05197	2.84677	1.35832	.054956		4	2.55994	2.38897	1.30255	.06595
	5	3.17212	2.90542	1.48108	.068287		5	2.64389	2.42442	1.40051	.08165
	6	3.30680	2.97131	1.62361	.081470		6	2.73534	2.46403	1.51078	.09752
	7	3.45826	3.04552	1.79138	.094530		7	2.85560	2.50808	1.63601	.11300
	8	3.63087	3.12950	1.99170	.10746		8	2.91639	2.55727	1.77961	.12830
	9	3.83033	3.22498	2.23535	.12029		9	3.06978	2.61226	1.94606	.14342
	10	4.06162	3.33411	2.53821	.13301		10	3.20843	2.61370	2.14123	.15838
	11	4.34571	3.45998	2.92528	.14564		11	3.36613	2.74282	2.37366	.17319
	12	4.69161	3.60650	3.43734	.15817		12	3.51771	2.82071	2.65513	.18786
	13	5.13319	3.77921	4.11788	.17065		13	3.76027	2.90907	3.00398	.20241
	14	5.72501	3.98579	5.19945	.18304		14	4.00005	2.99945	3.44555	.21835
	15	6.57835	4.23809	6.91836	.19538		15	4.32457	3.10529	4.02623	.23116
23	0	2.55931	2.55931	1.00000	0	27	0	2.20269	2.20269	1.00000	0
	1	2.63663	2.59191	1.07155	.014703		1	2.34209	2.30158	1.06314	.01614
	2	2.72021	2.62854	1.15131	.029212		2	2.40696	2.32466	1.13222	.03280
	3	2.81097	2.66964	1.24071	.043519		3	2.48322	2.35717	1.20820	.04823
	4	2.91011	2.71559	1.34175	.057649		4	2.55095	2.37983	1.29226	.06116
	5	3.01917	2.76708	1.45693	.071610		5	2.63139	2.41239	1.38571	.07958
	6	3.14004	2.82478	1.58952	.085414		6	2.71871	2.44871	1.49044	.09479
	7	3.27510	2.88954	1.74350	.090668		7	2.81393	2.48911	1.60857	.10980
	8	3.42766	2.95135	1.92598	.11259		8	2.91852	2.53112	1.74297	.12661
	9	3.60192	3.04481	2.14416	.12628		9	3.03424	2.58426	1.89742	.13925
	10	3.80388	3.13837	2.41057	.13927		10	3.16334	2.64029	2.07680	.15373
	11	4.04200	3.24533	2.74332	.15214		11	3.30871	2.70290	2.28776	.16804
	12	4.32897	3.36862	3.17120	.16551		12	3.47453	2.77291	2.53991	.18222
	13	4.68441	3.51192	3.74181	.17850		13	3.66599	2.85242	2.81643	.19627
	14	5.11181	3.68084	4.54240	.19141		14	3.89064	2.91427	3.22752	.21019
	15	5.76160	3.88284	5.74666	.20424		15	4.16067	3.04466	3.71441	.22400
24	0	2.45857	2.45857	1.00000	0	28	0	2.13006	2.13006	1.00000	0
	1	2.53038	2.48739	0.69615	.015428		1	2.18546	2.14729	1.06148	.01839
	2	2.60761	2.51974	1.14573	.030617		2	2.24436	2.16698	1.12856	.03646
	3	2.69120	2.55602	1.23122	.045607		3	2.30713	2.18917	1.20202	.05122
	4	2.78202	2.59660	1.32717	.060389		4	2.37434	2.21407	1.28293	.07170
	5	2.88137	2.64194	1.43578	.074985		5	2.44661	2.24198	1.37252	.08890
	6	2.99085	2.69267	1.55985	.089410		6	2.52465	2.27308	1.47228	.10585
	7	3.11233	2.71947	1.70294	.10367		7	2.60945	2.30767	1.50425	.12258
	8	3.24851	2.82309	1.86990	.11778		8	2.70205	2.34617	1.71071	.13906
	9	3.40126	2.88381	2.06753	.13183		9	2.89976	2.37065	1.85492	.14525
	10	3.57857	2.96549	2.30507	.14559		10	2.91678	2.43671	2.02095	.17143
	11	3.78354	3.05730	2.59633	.15930		11	3.04296	2.48992	2.21432	.18735
	12	4.02611	3.16209	2.96196	.17291		12	3.18538	2.54937	2.44239	.20308
	13	4.31993	3.28285	3.43525	.18612		13	3.34815	2.61606	2.71591	.21866
	14	4.68616	3.42314	4.07186	.19981		14	3.53684	2.69124	3.01990	.23410
	15	5.16169	3.58857	4.97574	.21317		15	3.75932	2.77640	3.46732	.24941

TABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO,
AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Continued

θ (deg)	β (deg)	M_b	M_a	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_0}$	θ (deg)	β (deg)	M_b	M_a	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_0}$
29	0	2.06266	2.06266	1.00000	0.	36	0	1.70130	1.70130	1.00000	0
	1	2.11505	2.07774	1.06001	.01910		1	1.73955	1.70539	1.05304	.025040
	2	2.17054	2.09196	1.12523	.03797		2	1.77947	1.71085	1.10969	.049148
	3	2.22977	2.11155	1.19647	.05616		3	1.82126	1.71772	1.17032	.073358
	4	2.29236	2.13521	1.27431	.07457		4	1.86536	1.72606	1.23585	.096650
	5	2.36005	2.16131	1.36067	.09251		5	1.91119	1.73654	1.30563	.11953
	6	2.43270	2.18895	1.45616	.11011	38	6	1.95980	1.74734	1.38147	.14189
	7	2.51112	2.21966	1.56258	.12744		7	2.01121	1.76042	1.46381	.16380
	8	2.59661	2.25386	1.68225	.14455		8	2.06585	1.77522	1.55356	.18530
	9	2.69398	2.29151	1.82347	.16209		9	2.12402	1.79185	1.65178	.20639
	10	2.79292	2.33402	1.97234	.17808		10	2.18626	1.81048	1.75992	.22713
30	11	2.90717	2.38101	2.15090	.19454	40	11	2.25305	1.83118	1.87943	.24749
	12	3.03508	2.43337	2.35935	.21081		12	2.32517	1.86418	2.01251	.26754
	13	3.17984	2.49183	2.50602	.22690		13	2.40336	1.87960	2.16155	.28728
	14	3.31576	2.55740	2.90292	.24285		14	2.48870	1.90778	2.32982	.30673
	15	3.55866	2.65123	3.26710	.25864		15	2.58245	1.93889	2.52146	.32591
31	0	2.00000	2.00000	1.00000	0	42	0	1.62127	1.62127	1.00000	0
	1	2.04970	2.01309	1.05869	.01996		1	1.66000	1.62610	1.05191	.02691
	2	2.10220	2.02817	1.2228	.03953		2	1.69720	1.62925	1.10711	.05312
	3	2.15786	2.04536	1.19144	.05873		3	1.73595	1.63363	1.16595	.07867
	4	2.21712	2.06489	1.26706	.07742		4	1.77646	1.63934	1.22888	.10361
	5	2.28038	2.08676	1.35004	.09616		5	1.81891	1.64639	1.29635	.12796
	6	2.34624	2.11124	1.44166	.11442	44	6	1.86348	1.65484	1.36892	.15177
	7	2.41213	2.13857	1.54331	.13239		7	1.91042	1.66171	1.44727	.17507
	8	2.50046	2.16931	1.65693	.15010		8	1.95999	1.67607	1.53211	.19788
	9	2.58664	2.20259	1.78180	.16757		9	2.03254	1.68903	1.62141	.22023
	10	2.68100	2.24016	1.92978	.18479		10	2.06839	1.70361	1.72524	.24217
32	11	2.78510	2.28175	2.09572	.20180	46	11	2.12800	1.71998	1.83585	.26369
	12	2.90088	2.32799	2.28775	.21861		12	2.19194	1.73827	1.95797	.28484
	13	3.03082	2.37952	2.51256	.23523		13	2.26069	1.75819	2.09337	.30562
	14	3.17820	2.43701	2.77947	.25167		14	2.33512	1.77819	2.24461	.32607
	15	3.34766	2.50119	3.10201	.26795		15	2.41604	1.80577	2.41463	.34621
33	0	1.88707	1.88707	1.00000	0	48	0	1.55572	1.55572	1.00000	0
	1	1.93213	1.89664	1.05659	.02158		1	1.58940	1.55556	1.05106	.02887
	2	1.97956	1.90799	1.11716	.04271		2	1.62431	1.55660	1.10515	.05693
	3	2.02958	1.92112	1.18287	.06342		3	1.66120	1.55877	1.16355	.08467
	4	2.08253	1.93620	1.25420	.08372		4	1.69811	1.56224	1.22384	.11085
	5	2.13775	1.95330	1.33195	.10367		5	1.73791	1.56694	1.28916	.13677
	6	2.19868	1.97256	1.41711	.12326	50	6	1.77912	1.57276	1.35914	.16209
	7	2.26277	2.00680	1.51078	.14251		7	1.82243	1.57993	1.43431	.18681
	8	2.33163	2.01807	1.61145	.16146		8	1.86797	1.58841	1.51532	.21098
	9	2.40576	2.04478	1.72982	.18011		9	1.91577	1.59826	1.60291	.23562
	10	2.48665	2.07445	1.85915	.19849		10	1.96679	1.60950	1.69799	.25777
34	11	2.57474	2.10730	2.00522	.21662	52	11	2.02072	1.62230	1.80165	.28046
	12	2.67149	2.11373	2.17150	.23150		12	2.07818	1.63666	1.91519	.30272
	13	2.77856	2.18409	2.36269	.25215		13	2.13960	1.65272	2.01007	.32156
	14	2.89812	2.22897	2.58506	.26960		14	2.20564	1.67062	2.17838	.34603
	15	3.03292	2.27884	2.81700	.28684		15	2.27681	1.69040	2.33224	.36713
35	0	1.78829	1.78829	1.03000	0	54	0	1.49448	1.49448	1.00000	0
	1	1.82960	1.79493	1.05153	.023271		1	1.52616	1.49254	1.05017	.03094
	2	1.87291	1.80209	1.11302	.046028		2	1.55923	1.49172	1.10379	.06096
	3	1.93837	1.81285	1.17591	.060285		3	1.59084	1.48920	1.16026	.09046
	4	1.96630	1.82127	1.24382	.090089		4	1.63013	1.49331	1.22140	.11902
	5	2.01688	1.83741	1.31735	.11115		5	1.66642	1.49592	1.28389	.14604
	6	2.07054	1.85237	1.39734	.13240	56	6	1.70602	1.50047	1.35186	.17270
	7	2.12758	1.86926	1.48470	.15297		7	1.74533	1.50437	1.42453	.19909
	8	2.18819	1.88821	1.58061	.17318		8	1.78759	1.51037	1.50251	.22462
	9	2.25576	1.90935	1.68638	.19304		9	1.83136	1.51758	1.58642	.24962
	10	2.32103	1.93285	1.80374	.21259		10	1.87871	1.52603	1.67703	.27402
36	11	2.40006	1.95894	1.93476	.23182	58	11	1.92809	1.53583	1.77522	.29790
	12	2.48279	1.96794	2.08213	.25078		12	1.98043	1.54697	1.88207	.31218
	13	2.57339	2.01986	2.24923	.26948		13	2.03608	1.55958	1.99882	.31119
	14	2.67317	2.05523	2.44021	.28792		14	2.09550	1.57370	2.12706	.36667
	15	2.78406	2.07055	2.66102	.30614		15	2.15919	1.58941	2.26861	.38873

TABLE I. - VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO,
AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Continued.

θ (deg)	β (deg)	M_b	M_a	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_0}$	θ (deg)	β (deg)	M_b	M_a	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_0}$
44	0	1.43955	1.43955	1.00000	0	52	0	1.26902	1.26902	1.00000	0
	1	1.47017	1.43595	1.05016	.03515		1	1.29670	1.25948	1.05144	.04370
	2	1.50175	1.43310	1.10300	.06524		2	1.32498	1.25103	1.10516	.08557
	3	1.53129	1.43131	1.15878	.09634		3	1.35392	1.24364	1.16131	.12571
	4	1.56816	1.43151	1.21777	.12651		4	1.38559	1.23720	1.22017	.16450
	5	1.60320	1.43215	1.28033	.15581		5	1.41404	1.23173	1.28189	.20140
	6	1.63960	1.43385	1.34680	.18129		6	1.44153	1.22720	1.34680	.23715
	7	1.67754	1.43559	1.41765	.21202		7	1.47766	1.22354	1.41517	.27165
	8	1.71712	1.44014	1.49329	.23900		8	1.51097	1.22079	1.48728	.30491
	9	1.75855	1.44526	1.57835	.26532		9	1.54543	1.21981	1.56357	.33709
	10	1.80261	1.45143	1.66148	.29101		10	1.58114	1.21788	1.64446	.36826
	11	1.84775	1.45668	1.75542	.31608		11	1.61819	1.21844	1.73036	.39845
	12	1.89597	1.46707	1.85710	.34062		12	1.65677	1.21836	1.82188	.42775
	13	1.94702	1.47675	1.96754	.36661		13	1.69700	1.21989	1.91963	.45620
14	2.00123	1.48774	2.08604	.37474	↓	14	1.72905	1.22229	2.02439	.48384	
	15	2.05902	1.50014	2.22011	.41113	15	1.76314	1.22557	2.13679	.51075	
46	0	1.39016	1.39016	1.00000	0	54	0	1.23606	1.23606	1.00000	0
	1	1.41169	1.38098	1.05010	.03551		1	1.26353	1.22509	1.05243	.04691
	2	1.45008	1.38088	1.10274	.06980		2	1.29155	1.21520	1.10710	.09172
	3	1.48138	1.37779	1.15813	.10294		3	1.32016	1.20650	1.16416	.13456
	4	1.51372	1.37973	1.21661	.13505		4	1.36194	1.19873	1.22383	.17560
	5	1.54715	1.37164	1.27839	.16615		5	1.37942	1.19194	1.28631	.21495
	6	1.58181	1.37160	1.34384	.19631		6	1.41018	1.18607	1.35186	.25277
	7	1.61778	1.37510	1.41331	.22562		7	1.44181	1.18109	1.42075	.28914
	8	1.65521	1.37744	1.48728	.25408		8	1.47440	1.17701	1.49329	.32417
	9	1.69426	1.38037	1.56626	.28181		9	1.50801	1.17379	1.56982	.35796
	10	1.73507	1.38451	1.65076	.30881		10	1.54274	1.17135	1.65074	.39059
	11	1.77783	1.38936	1.74114	.33512		11	1.57871	1.16978	1.73640	.42214
	12	1.82279	1.39515	1.83915	.36080		12	1.61605	1.16899	1.82758	.45269
	13	1.87015	1.40266	1.94474	.38589		13	1.65487	1.16904	1.92455	.48229
14	1.92023	1.41102	2.05931	.41061	↓	14	1.69536	1.17067	2.02809	.51099	
	15	1.97330	1.42055	2.18408	.43441	15	1.73765	1.17155	2.13897	.53888	
48	0	1.34562	1.34562	1.00000	0	56	0	1.20621	1.20621	1.00000	0
	1	1.37131	1.33897	1.05028	.03803		1	1.23369	1.19378	1.05375	.05045
	2	1.40576	1.35334	1.10299	.07466		2	1.26164	1.18252	1.10269	.09845
	3	1.43404	1.32876	1.15835	.11000		3	1.29013	1.17238	1.16799	.14119
	4	1.46522	1.32515	1.21660	.14113		4	1.31923	1.16331	1.22687	.18787
	5	1.49738	1.32253	1.27799	.17712		5	1.34896	1.15521	1.29250	.22963
	6	1.53062	1.32084	1.34285	.20906		6	1.37943	1.14807	1.35916	.26964
	7	1.56167	1.32016	1.41075	.23968		7	1.41067	1.14184	1.42906	.30801
	8	1.60075	1.32034	1.48134	.27003		8	1.44280	1.13650	1.50253	.34487
	9	1.63789	1.32151	1.56195	.29919		9	1.47583	1.13198	1.57987	.38033
	10	1.67656	1.32364	1.64444	.32752		10	1.50992	1.12828	1.66148	.41449
	11	1.71700	1.32672	1.73283	.35511		11	1.54513	1.12540	1.74774	.44743
	12	1.75929	1.33077	1.82758	.38198		12	1.58159	1.12331	1.82924	.47924
	13	1.80370	1.32585	1.92551	.40816		13	1.61925	1.12200	1.93582	.50988
14	1.85014	1.34190	2.03959	.43372	↓	14	1.65674	1.12114	2.03961	.53978	
	15	1.89980	1.34908	2.15885	.45868	15	1.69971	1.12167	2.14992	.56862	
50	0	1.30541	1.30541	1.00000	0	58	0	1.17918	1.17918	1.00000	0
	1	1.33318	1.29731	1.05072	.04075		1	1.20686	1.16522	1.05542	.05436
	2	1.36225	1.29028	1.10380	.07991		2	1.23498	1.15255	1.11302	.10586
	3	1.39174	1.28125	1.15940	.11756		3	1.26357	1.14102	1.17297	.15476
	4	1.42204	1.27920	1.21777	.15361		4	1.29271	1.13061	1.23547	.20130
	5	1.45322	1.27510	1.27916	.18884		5	1.32244	1.12122	1.30071	.24564
	6	1.48536	1.27194	1.34383	.22263		6	1.35284	1.11285	1.36892	.28796
	7	1.51856	1.26972	1.41210	.25529		7	1.38524	1.10526	1.44036	.32846
	8	1.55292	1.26837	1.48136	.28693		8	1.41585	1.09881	1.51533	.36724
	9	1.58854	1.26795	1.56095	.31756		9	1.44663	1.09309	1.59409	.40443
	10	1.62554	1.26645	1.64237	.34729		10	1.48236	1.08819	1.67705	.44017
	11	1.66408	1.27385	1.72917	.37617		11	1.51713	1.08408	1.76457	.47454
	12	1.70126	1.27204	1.82185	.40422		12	1.55305	1.08076	1.85710	.50765
	13	1.74634	1.27525	1.92125	.43154		13	1.59024	1.07621	1.95517	.53958
14	1.79047	1.27937	2.02809	.45614	↓	14	1.62880	1.07640	2.05932	.57042	
	15	1.83687	1.27004	2.14331	.48407	15	1.66888	1.07531	2.17022	.60023	

TABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO,
AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Concluded

θ (deg)	β (deg)	M_b	M_a	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_0}$	θ (deg)	β (deg)	M_b	M_a	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_0}$
60	0	1.15470	1.15470	1.00000	0	66	0	1.09463	1.09463	1.00000	0
	1	1.16281	1.13913	1.05748	.05869		1	1.12559	1.07321	1.06691	.07545
	2	1.21128	1.12492	1.11715	.11107		2	1.15678	1.05388	1.13626	.14547
	3	1.21021	1.11200	1.17919	.16645		3	1.18829	1.03613	1.20818	.21062
	4	1.26963	1.10024	1.24379	.21606		4	1.21761	1.01850	1.27689	.26680
	5	1.29959	1.08956	1.31115	.26318		5	1.25245	1.00511	1.36067	.32847
	6	1.33016	1.07991	1.38199	.30802		6	1.28523	.99157	1.44166	.38197
	7	1.36159	1.07123	1.45505	.35075		7	1.31856	.97920	1.52616	.42233
	8	1.39336	1.06349	1.52211	.39149		8	1.35250	.96785	1.61445	.47986
	9	1.42614	1.05661	1.61296	.43053		9	1.38715	.95764	1.70686	.53374
	10	1.45981	1.05056	1.69801	.46792		10	1.42256	.94829	1.80374	.56738
	11	1.49146	1.04530	1.78756	.50375		11	1.45577	.94000	1.89687	.60453
	12	1.53015	1.04083	1.88204	.53817		12	1.49504	.93219	2.01255	.61630
	13	1.56705	1.03711	1.98204	.57130		13	1.53126	.92532	2.12527	.68296
	14	1.60524	1.03411	2.08804	.60321		14	1.57368	.92268	2.24461	.71796
	15	1.64485	1.03183	2.20066	.63397		15	1.61143	.91388	2.37080	.75142
62	0	1.13257	1.13257	1.03000	0	68	0	1.07854	1.07854	1.00000	0
	1	1.16134	1.11527	1.06001	.06356		1	1.11113	1.05481	1.07157	.08281
	2	1.19044	1.09445	1.12228	.12326		2	1.14392	1.03325	1.1574	.15911
	3	1.21994	1.08503	1.18693	.17943		3	1.17698	1.01361	1.22268	.22961
	4	1.24988	1.07187	1.25120	.23246		4	1.21035	.99570	1.30259	.29508
	5	1.28033	1.05987	1.32125	.28258		5	1.24412	.97935	1.38571	.35599
	6	1.31132	1.04897	1.39734	.33010		6	1.27834	.96440	1.47229	.41287
	7	1.34294	1.03907	1.47368	.37521		7	1.31308	.95070	1.56259	.46613
	8	1.37527	1.03015	1.55357	.41831		8	1.34843	.93821	1.65696	.51616
	9	1.40834	1.02212	1.63750	.45902		9	1.38446	.92678	1.75568	.56322
	10	1.44122	1.01497	1.72524	.49808		10	1.42122	.91635	1.85916	.60756
	11	1.48051	1.01097	1.81770	.53293		11	1.45885	.90685	1.96784	.64966
	12	1.51292	1.00303	1.91519	.57190		12	1.49741	.89825	2.08216	.6895
	13	1.54988	.99818	2.01813	.60549		13	1.53698	.89042	2.20262	.72727
	14	1.58806	.99116	2.12710	.63846		14	1.57772	.88338	2.32987	.76322
	15	1.62757	.99077	2.24268	.67016		15	1.61972	.87707	2.46455	.79749
64	0	1.11260	1.11260	1.00000	0	70	0	1.06418	1.06418	1.00000	0
	1	1.14230	1.09339	1.06311	.06909		1	1.09890	1.03765	1.07737	.09153
	2	1.17229	1.07586	1.12854	.13362		2	1.13406	1.01384	1.15827	.17580
	3	1.20263	1.05981	1.19846	.19405		3	1.16883	.99194	1.24074	.25174
	4	1.23329	1.04513	1.26705	.25082		4	1.20418	.97213	1.32717	.32233
	5	1.26456	1.03183	1.34052	.30420		5	1.23990	.95409	1.41710	.38759
	6	1.29631	1.01963	1.41710	.35459		6	1.27606	.93761	1.51080	.48114
	7	1.32862	1.00852	1.49702	.40223		7	1.31270	.92100	1.60854	.50450
	8	1.36159	.99814	1.56060	.44739		8	1.34994	.90880	1.71071	.55714
	9	1.39526	.98927	1.66809	.49026		9	1.38786	.89621	1.81765	.60643
	10	1.42974	.98100	1.75989	.53105		10	1.42754	.88510	1.93274	.65386
	11	1.46509	.97358	1.85635	.56993		11	1.46606	.87417	2.04755	.69626
	12	1.50112	.96697	1.95793	.60706		12	1.50653	.86458	2.17117	.73736
	13	1.53881	.96113	2.05507	.61255		13	1.54805	.85582	2.30216	.77624
	14	1.57739	.95602	2.17835	.67655		14	1.59076	.84768	2.44024	.81307
	15	1.61725	.95161	2.29836	.70916		15	1.63475	.84069	2.58643	.84805

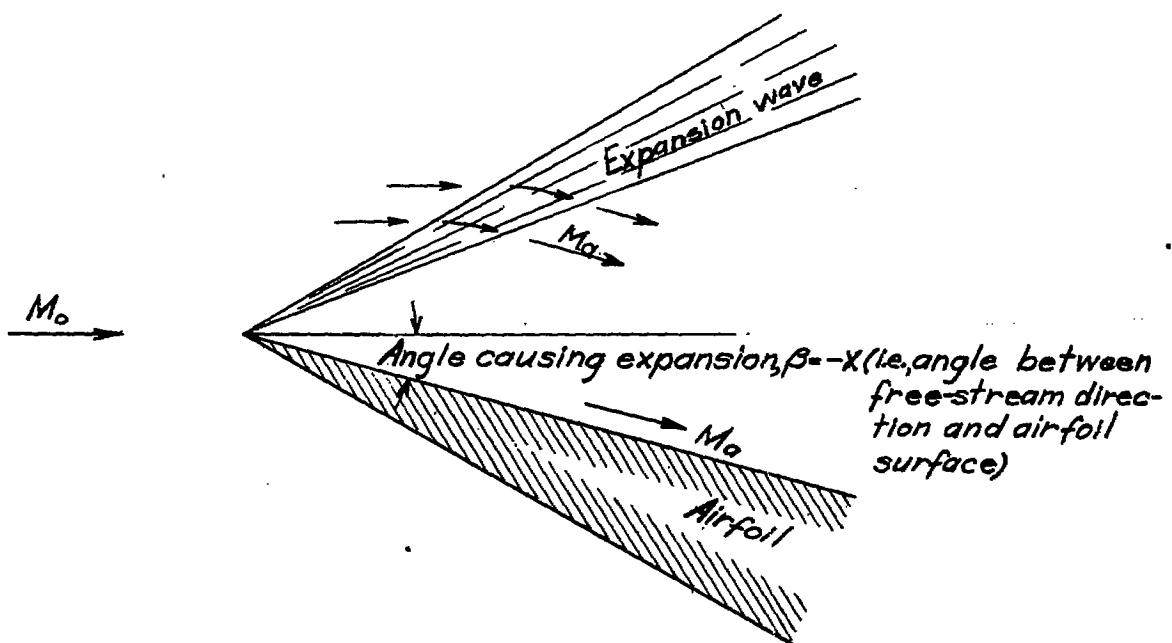
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TABLE IV.- PRESSURE COEFFICIENT BASED ON FREE-STREAM DYNAMIC PRESSURE

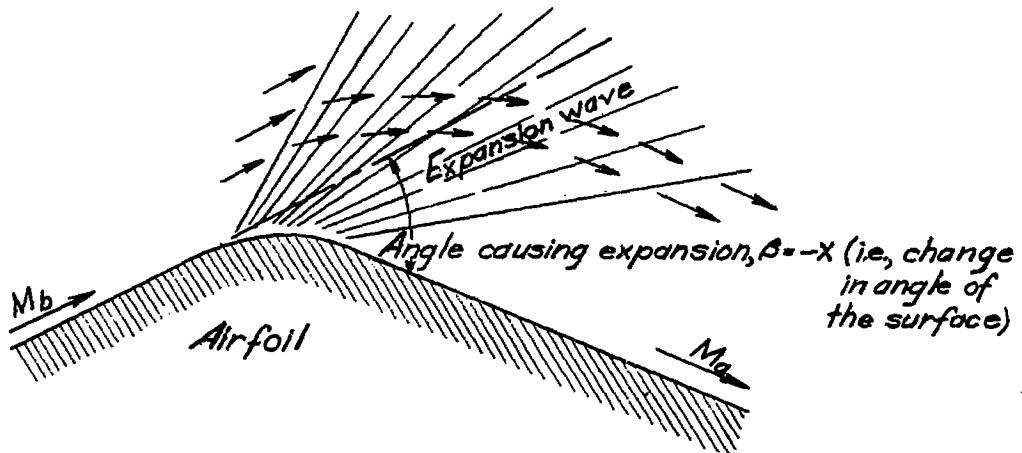
		Pressure coefficient, $\Delta p_d/q_\infty$							
		1.0	1.25	1.5	1.75	2.0	2.5	3.0	3.5
M_∞	P_d/P_∞								
0	-1.42857	-0.91429	-0.63492	-0.47579	-0.35714	-0.22857	-0.15873	-0.11662	
.05	-1.35714	-0.86357	-0.60317	-0.45200	-0.33929	-0.21714	-0.15079	-0.11079	
.1	-1.28571	-0.82286	-0.57143	-0.42821	-0.32113	-0.20571	-0.14286	-0.10496	
.2	-1.14286	-0.73143	-0.50794	-0.38064	-0.28571	-0.18286	-0.12698	-0.095294	
.3	-1.00000	-0.61000	-0.44444	-0.33306	-0.25000	-0.16000	-0.11111	-0.081633	
.4	-0.85714	-0.51487	-0.38092	-0.28548	-0.21429	-0.13714	-0.095238	-0.069971	
.5	-0.71429	-0.45714	-0.31746	-0.23790	-0.17857	-0.11429	-0.079365	-0.058309	
.6	-0.57143	-0.36571	-0.25397	-0.19032	-0.14286	-0.091129	-0.063492	-0.046647	
.7	-0.42857	-0.27429	-0.19048	-0.14274	-0.10714	-0.068571	-0.047619	-0.034985	
.8	-0.28571	-0.18286	-0.12698	-0.095159	-0.071429	-0.045714	-0.031746	-0.023321	
.9	-0.14286	-0.091429	-0.063492	-0.047579	-0.035714	-0.022857	-0.015873	-0.011662	
1.0	0	0	0	0	0	0	0	0	
1.1	.114286	.091429	.063492	.047579	.035714	.022857	.015873	.011662	
1.2	.28571	.18286	.12698	.095159	.071429	.05714	.031746	.023321	
1.3	.42857	.27429	.19048	.14274	.10714	.068571	.047619	.034985	
1.4	.57143	.36571	.25397	.19052	.14286	.091129	.063492	.046647	
1.5	.71429	.45714	.31746	.23790	.17857	.11429	.079365	.058309	
1.6	.85714	.54857	.38095	.28418	.21429	.13714	.095238	.069971	
1.7	1.00000	.61000	.44444	.33306	.25000	.16000	.11111	.081633	
1.8	1.14286	.73143	.20794	.38064	.28571	.18286	.12698	.095294	
1.9	1.28571	.82286	.57113	.42821	.32143	.20571	.14286	.10496	
2.0	1.42857	.91429	.63492	.47579	.35714	.22857	.15873	.11662	
2.2	1.71429	1.09714	.76190	.57095	.42857	.27429	.19048	.13994	
2.4	2.00000	1.28000	.88889	.66611	.50000	.32000	.22222	.16327	
2.6	2.28571	1.46286	1.01587	.76127	.57143	.42857	.30991	.20991	
2.8	2.57143	1.64571	1.14286	.85643	.64286	.44143	.31746	.23324	
3.0	2.85714	1.82957	1.26984	.95159	.71429	.45714	.34921	.25656	
3.2	3.14286	2.01143	1.39683	1.06175	.78571	.50286	.38095	.27988	
3.4	3.42857	2.19129	1.52381	1.14191	.85714	.54857	.41270	.30521	
3.6	3.71429	2.37714	1.65979	1.23706	.92857	.59429	.44515	.32652	
3.8	4.00000	2.56000	1.77778	1.33222	1.00000	.64000	.47619	.34985	
4.0	4.28571	2.74286	1.90476	1.42738	1.07143	.68571			
4.2	4.57143	2.92571	2.07175	1.52254	1.14286	.73143	.50794	.37318	
4.4	4.85714	3.10857	2.15875	1.61770	1.21429	.77714	.53968	.39650	
4.6	5.14286	3.29243	2.28571	1.71286	1.28571	.82286	.57113	.41983	
4.8	5.42857	3.47429	2.11270	1.80802	1.35714	.86857	.60517	.44515	
5.0	5.71429	3.65714	2.53968	1.90318	1.42657	.91429	.63492	.46647	
5.2	6.00000	3.84000	2.66667	1.99833	1.50000	.96000	.66667	.48980	
5.4	6.28511	4.02286	2.79365	2.09319	1.57113	1.00571	.69841	.51312	
5.6	6.57143	4.20571	2.92063	2.18862	1.64286	1.05143	.73016	.53644	
5.8	6.85714	4.38857	3.04762	2.28381	1.71429	1.09714	.76190	.55977	
6.0	7.14286	4.57143	3.17460	2.37897	1.78571	1.14286	.79365	.58309	
6.2	7.42857	4.75429	3.30159	2.47415	1.85714	1.18857	.82540	.60641	
6.4	7.71429	4.93714	3.42857	2.56929	1.92857	1.23129	.85714	.62974	
6.6	8.00000	5.12000	3.55556	2.66145	2.00000	1.28000	.88889	.65306	
6.8	8.28571	5.30286	3.68254	2.75960	2.07143	1.32571	.92062	.67638	
7.0	8.57143	5.48957	3.80952	2.85476	2.14286	1.37143	.95238	.69971	
7.2	8.85714	5.66857	3.93651	2.94992	2.21429	1.41714	.98413	.72303	
7.4	9.14286	5.85143	4.06349	3.04508	2.28571	1.46286	1.01587	.74636	
7.6	9.42857	6.05428	4.19048	3.14024	2.35714	1.50857	1.04762	.76968	
7.8	9.71429	6.21714	4.31746	2.35540	2.42857	1.55429	1.07936	.79300	
8.0	10.00000	6.40000	4.44444	3.33056	2.50000	1.60000	1.11111	.81633	
8.2	10.28571	6.58286	4.57143	3.42572	2.57143	1.64571	1.14286	.83965	
8.4	10.57143	6.76571	4.69811	3.52087	2.61286	1.69143	1.17460	.86297	
8.6	10.85714	6.94857	4.82511	3.61603	2.71429	1.75714	1.20635	.88630	
8.8	11.14286	7.13143	4.95238	3.71119	2.78571	1.78286	1.23809	.90962	
9.0	11.42857	7.31428	5.07936	3.80635	2.85714	1.82857	1.26984	.93294	
9.2	11.71429	7.49714	5.20635	3.90151	2.92857	1.87429	1.30159	.95627	
9.4	12.00000	7.68000	5.33333	3.99667	3.00000	1.92000	1.33333	.97959	
9.6	12.28571	7.86286	5.46032	4.09183	3.07143	1.96571	1.36508	1.00292	
9.8	12.57143	8.04571	5.58750	4.18699	3.11286	2.01143	1.39682	1.02624	
10.0	12.85714	8.22857	5.71429	4.28215	3.21429	2.05714	1.42857	1.04956	

TABLE IV.- PRESSURE COEFFICIENT BASED ON FREE-STREAM DYNAMIC PRESSURE - Concluded

		Pressure coefficient, $\Delta P_{\infty}/q_0$								
M_∞	P_n/P_0	4.0	4.5	5.0	5.5	6.0	6.5	7.0	7.5	8.0
0	-0.089286	-0.070547	-0.057143	-0.047226	-0.039683	-0.033812	-0.029155	-0.025397	-0.022321	-0.021205
.05	-0.084821	-0.067019	-0.054286	-0.048641	-0.037698	-0.032122	-0.027697	-0.024127	-0.022857	-0.020089
.1	-0.080357	-0.063192	-0.051129	-0.042503	-0.035714	-0.030431	-0.026239	-0.022857	-0.020317	-0.017857
.2	-0.0711429	-0.056427	-0.045714	-0.037780	-0.031746	-0.027050	-0.023524	-0.020317	-0.017778	-0.015625
.3	-0.062500	-0.049382	-0.040000	-0.033058	-0.027778	-0.023669	-0.020108	-0.017193	-0.015238	-0.013393
.4	-0.053571	-0.042328	-0.034286	-0.028352	-0.023610	-0.020287	-0.017193	-0.015238	-0.013393	-0.011161
.5	-0.044643	-0.035273	-0.028571	-0.023613	-0.019811	-0.016906	-0.014577	-0.012698	-0.011161	-0.011161
.6	-0.035714	-0.028219	-0.022857	-0.018890	-0.015873	-0.013525	-0.011662	-0.010159	-0.0089286	-0.006964
.7	-0.026786	-0.021164	-0.017142	-0.01168	-0.011905	-0.010144	-0.0087164	-0.0076190	-0.0050794	-0.0041643
.8	-0.017857	-0.011109	-0.011429	-0.0091451	-0.0079365	-0.0067625	-0.0053309	-0.00425397	-0.0022321	-0.0022321
.9	-0.0089286	-0.0070547	-0.0057143	-0.0047226	-0.0039683	-0.0033812	-0.0029155	0	0	0
1.0	0	0	0	0	0	0	0	0	0	0
1.1	0.0089286	0.0070547	0.0057143	0.0047226	0.0039683	0.0033812	0.0029155	0.0025397	0.0022321	0.0022321
1.2	0.017857	0.014169	0.011429	0.0091451	0.0079365	0.0067625	0.0053309	0.0050794	0.0041643	0.0035714
1.3	0.026786	0.021164	0.017142	0.01168	0.011905	0.010144	0.0087164	0.0076190	0.0066964	0.0066964
1.4	0.035714	0.028219	0.022857	0.018890	0.015873	0.013525	0.011662	0.010159	0.0089286	0.0089286
1.5	0.044643	0.035273	0.028571	0.023613	0.019811	0.016906	0.014577	0.012698	0.011161	0.011161
1.6	0.053571	0.042328	0.034286	0.028352	0.023810	0.020287	0.017493	0.015238	0.013393	0.013393
1.7	0.062500	0.049383	0.040000	0.033058	0.027778	0.023669	0.020408	0.017778	0.015625	0.015625
1.8	0.071129	0.056437	0.051714	0.037780	0.031746	0.027050	0.023324	0.020317	0.017857	0.017857
1.9	0.080357	0.063492	0.051129	0.042503	0.035714	0.030131	0.026239	0.022857	0.020089	0.020089
2.0	0.089286	0.070547	0.057143	0.047226	0.039683	0.033812	0.029155	0.025397	0.022321	0.022321
2.2	.10714	.084656	.068571	.056671	.047619	.040575	.034985	.030476	.026786	.026786
2.4	.12500	.098765	.080000	.066116	.055556	.047337	.040816	.035556	.031250	.031250
2.6	.14286	.11287	.091129	.075561	.063542	.054100	.046647	.040635	.035714	.035714
2.8	.16071	.12698	.10286	.085006	.071429	.060862	.052478	.045714	.040179	.040179
3.0	.17857	.14109	.11429	.094451	.079365	.067625	.058509	.050794	.044643	.044643
3.2	.19653	.15520	.12571	.10390	.087302	.074387	.061140	.055873	.049107	.049107
3.4	.21429	.16931	.13714	.11334	.095238	.081150	.069971	.060952	.053571	.053571
3.6	.23214	.18342	.14857	.12279	.10317	.087912	.075802	.066032	.058036	.058036
3.8	.25000	.19753	.16000	.13223	.11111	.094674	.081633	.071111	.062500	.062500
4.0	.26786	.21164	.17143	.14168	.11905	.101044	.087464	.076190	.066964	.066964
4.2	.28571	.22575	.18286	.15112	.12698	.10820	.093294	.081270	.071129	.071129
4.4	.30357	.23986	.19429	.16057	.13492	.11496	.099125	.086349	.075893	.075893
4.6	.32143	.25297	.20571	.17901	.14286	.12172	.10494	.091129	.080357	.080357
4.8	.33929	.26808	.21714	.17946	.15079	.12849	.11075	.096508	.081821	.081821
5.0	.35714	.28219	.22857	.18890	.15873	.13525	.11656	.10159	.089286	.089286
5.2	.37500	.29630	.24000	.19835	.16667	.14201	.12237	.10667	.093750	.093750
5.4	.39286	.31041	.25113	.20779	.17460	.15877	.12818	.11175	.098214	.098214
5.6	.41071	.32151	.26286	.21724	.18254	.15554	.13399	.11683	.10268	.10268
5.8	.42857	.33862	.27429	.22668	.19048	.16230	.13980	.12190	.10714	.10714
6.0	.44643	.35273	.28571	.23613	.19841	.16906	.14561	.12698	.11161	.11161
6.2	.46429	.36684	.29714	.24557	.20635	.17582	.151142	.13206	.11607	.11607
6.4	.48214	.38095	.30857	.25502	.21429	.18529	.15723	.13714	.12054	.12054
6.6	.50000	.39506	.32000	.26446	.22222	.18935	.16305	.14222	.12500	.12500
6.8	.51786	.40907	.33143	.27391	.23016	.19611	.16886	.14730	.12916	.12916
7.0	.53571	.42508	.34286	.28335	.23810	.20287	.17467	.15238	.13393	.13393
7.2	.55357	.43709	.35429	.29280	.24603	.20964	.18048	.15746	.13839	.13839
7.4	.57113	.45110	.36571	.30224	.25397	.21640	.18629	.16254	.14286	.14286
7.6	.58929	.46511	.37714	.31169	.26190	.22316	.19210	.16762	.14732	.14732
7.8	.60714	.47912	.38857	.32113	.26984	.22992	.19791	.17270	.15178	.15178
8.0	.62500	.49313	.40000	.33058	.27778	.23669	.20372	.17778	.15625	.15625
8.2	.64286	.50714	.41143	.34002	.28571	.24345	.20953	.18286	.16071	.16071
8.4	.66071	.52115	.42286	.34917	.29365	.25021	.21534	.18794	.16518	.16518
8.6	.67857	.53513	.43129	.35891	.30159	.25697	.22115	.19502	.16964	.16964
8.8	.69613	.54916	.44571	.36836	.30952	.26374	.22697	.19810	.17411	.17411
9.0	.71429	.56317	.45714	.37780	.31746	.27050	.23278	.20317	.17857	.17857
9.2	.73214	.57718	.46857	.38725	.32540	.27726	.23859	.20825	.18304	.18304
9.4	.75000	.59119	.48000	.39869	.33333	.28402	.24440	.21333	.18750	.18750
9.6	.76786	.60520	.49112	.40614	.34127	.29079	.25021	.21811	.19196	.19196
9.8	.78571	.61921	.50284	.41558	.34921	.29755	.25602	.22319	.19643	.19643
10.0	.80357	.63322	.51426	.42503	.35714	.30431	.26183	.22857	.20089	.20089



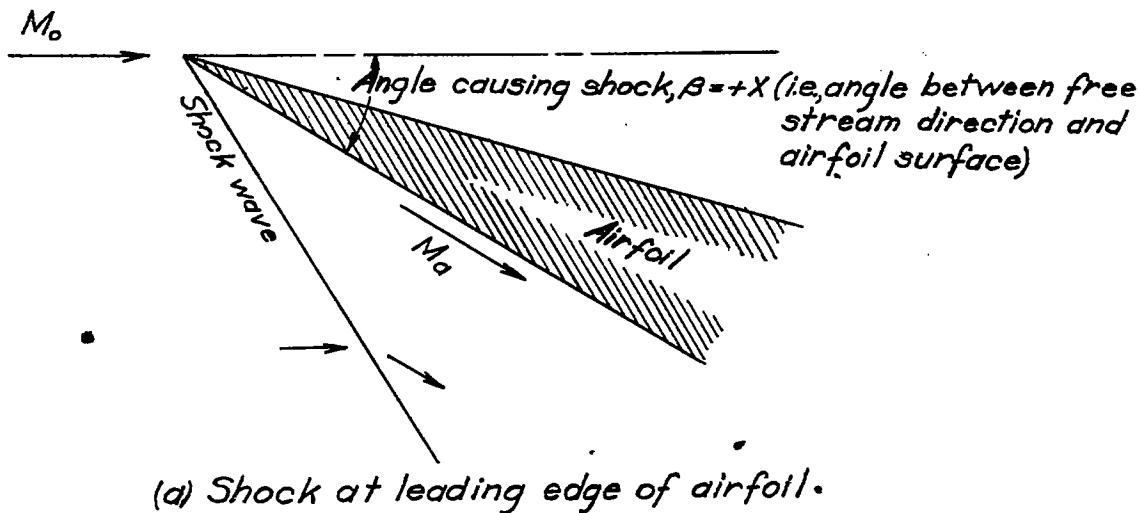
(a) Expansion at leading edge of airfoil.



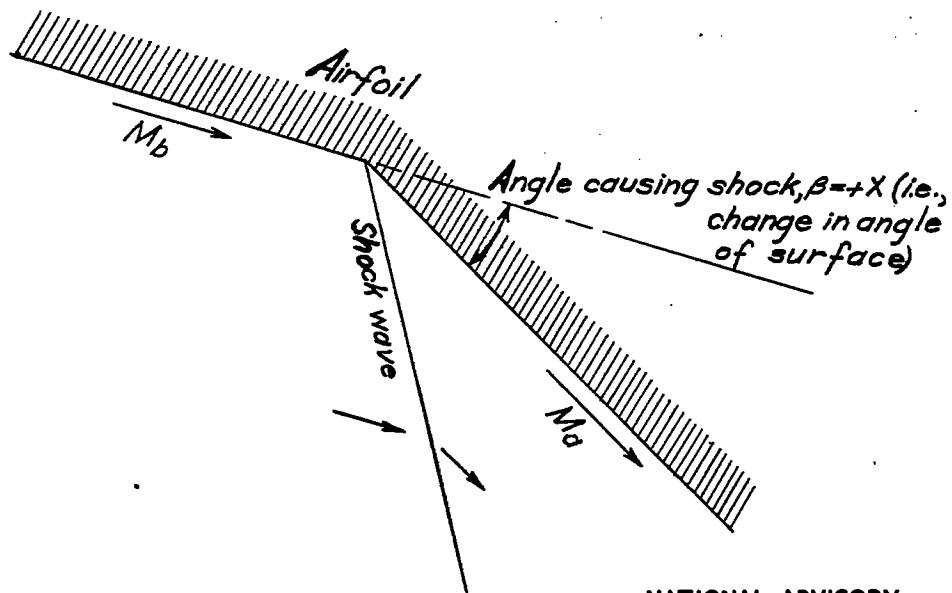
(b) Expansion along the airfoil.

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Figure 1.- Method of measuring angle causing expansion.
The angle causing expansion is always considered negative.



(a) Shock at leading edge of airfoil.

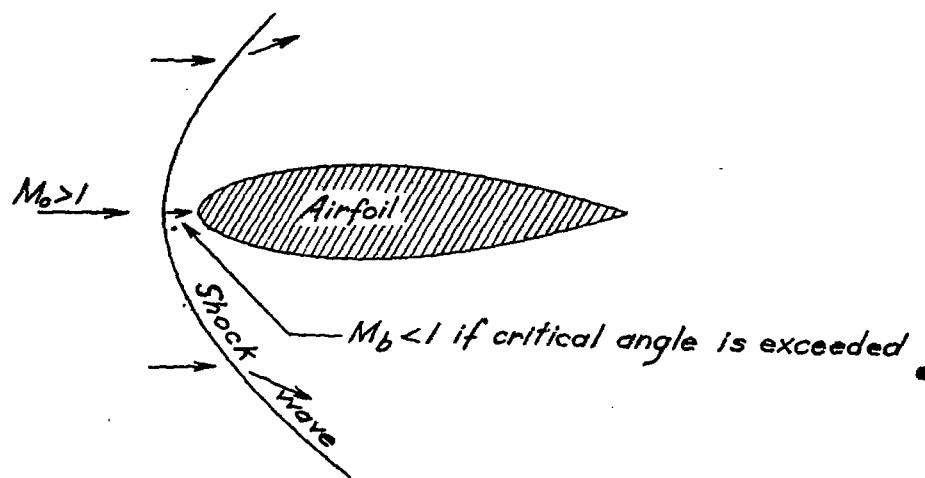
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(b) Shock at intermediate point along airfoil.

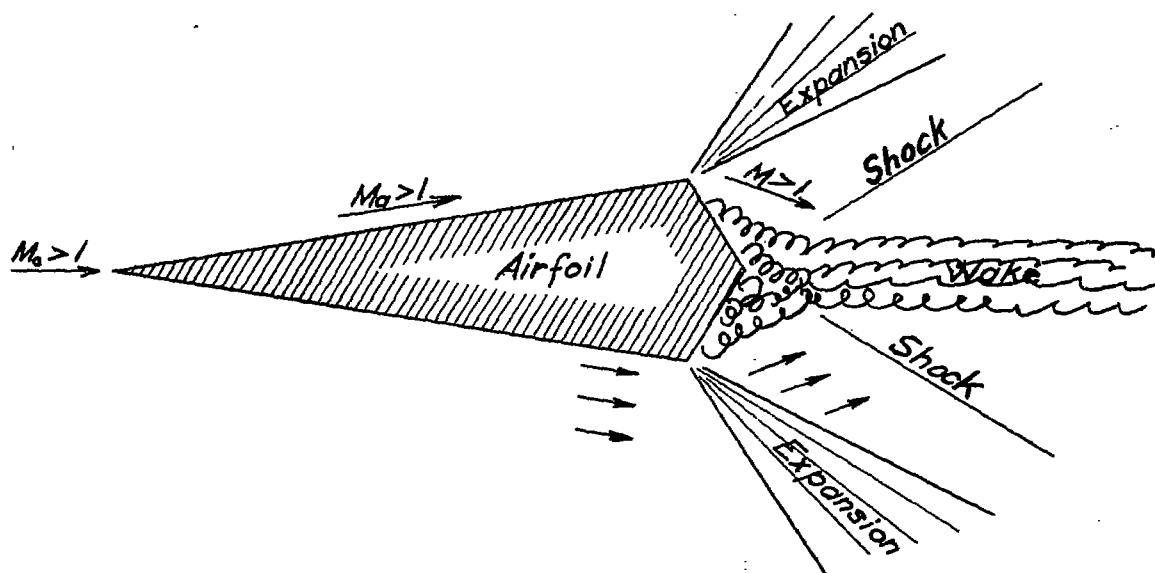
Figure 2.- Method of measuring angle causing shock. The angle causing shock is always considered positive.

Fig. 3a,b

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(a) Shock limitation exceeded.



(b) Expansion limitations exceeded (Turbulent wake set up.)

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Figure 3.- Effect of exceeding the limitation on angles.

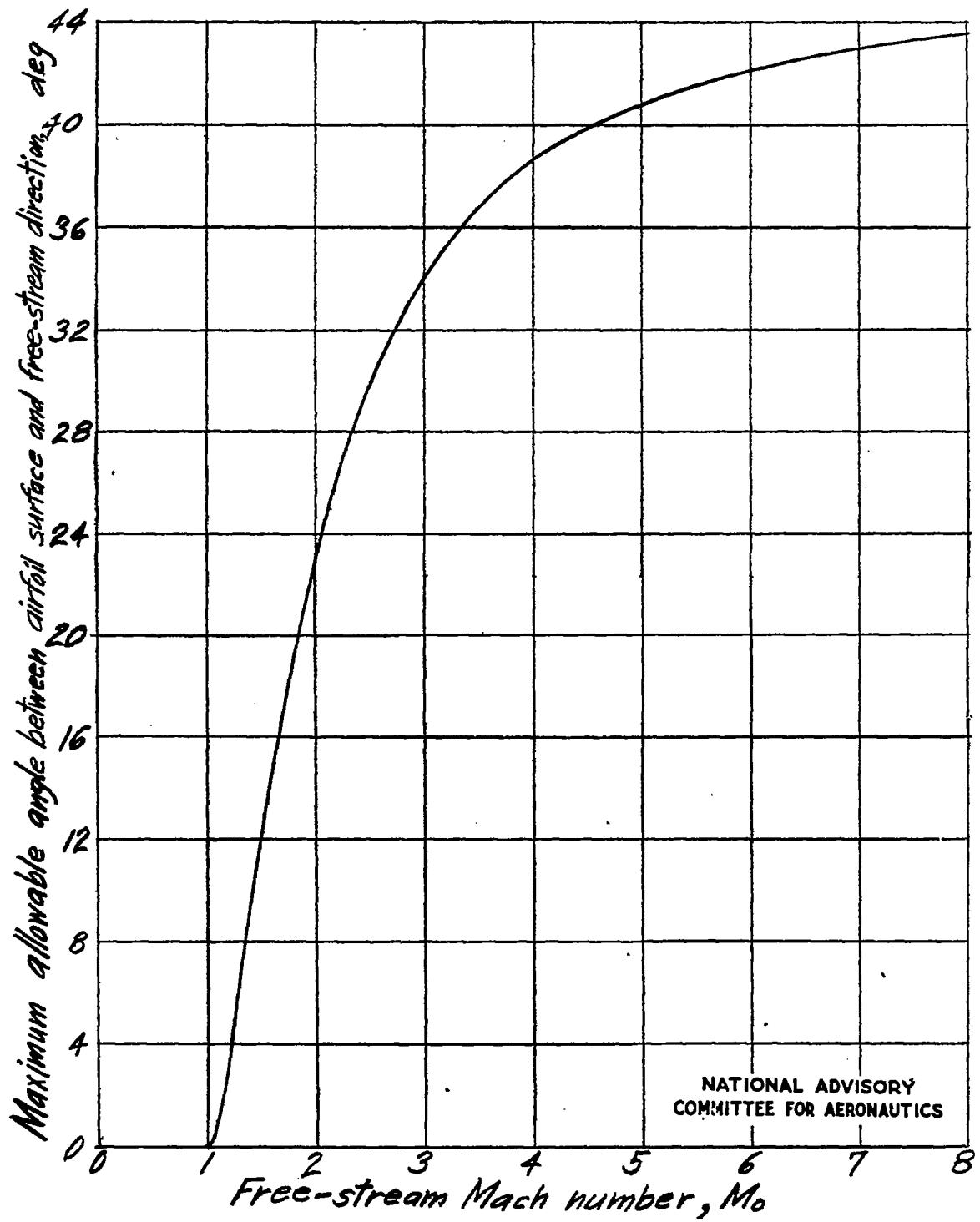


Figure 4.- Surface-angle limitation for attached shock wave.

Fig. 5

NACA TN No. 1143

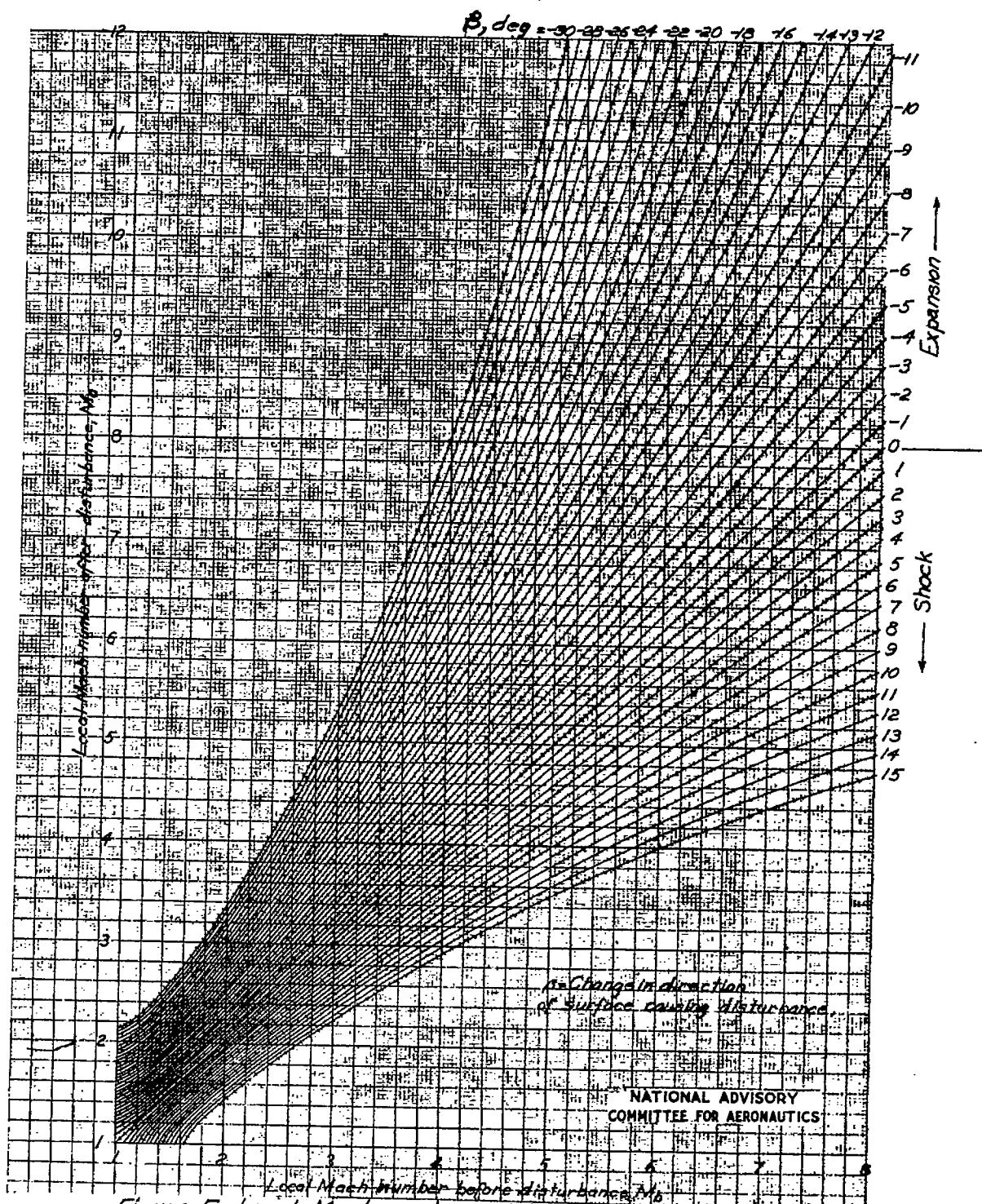


Figure 5.- Local Mach numbers before and after shocks
and expansions.

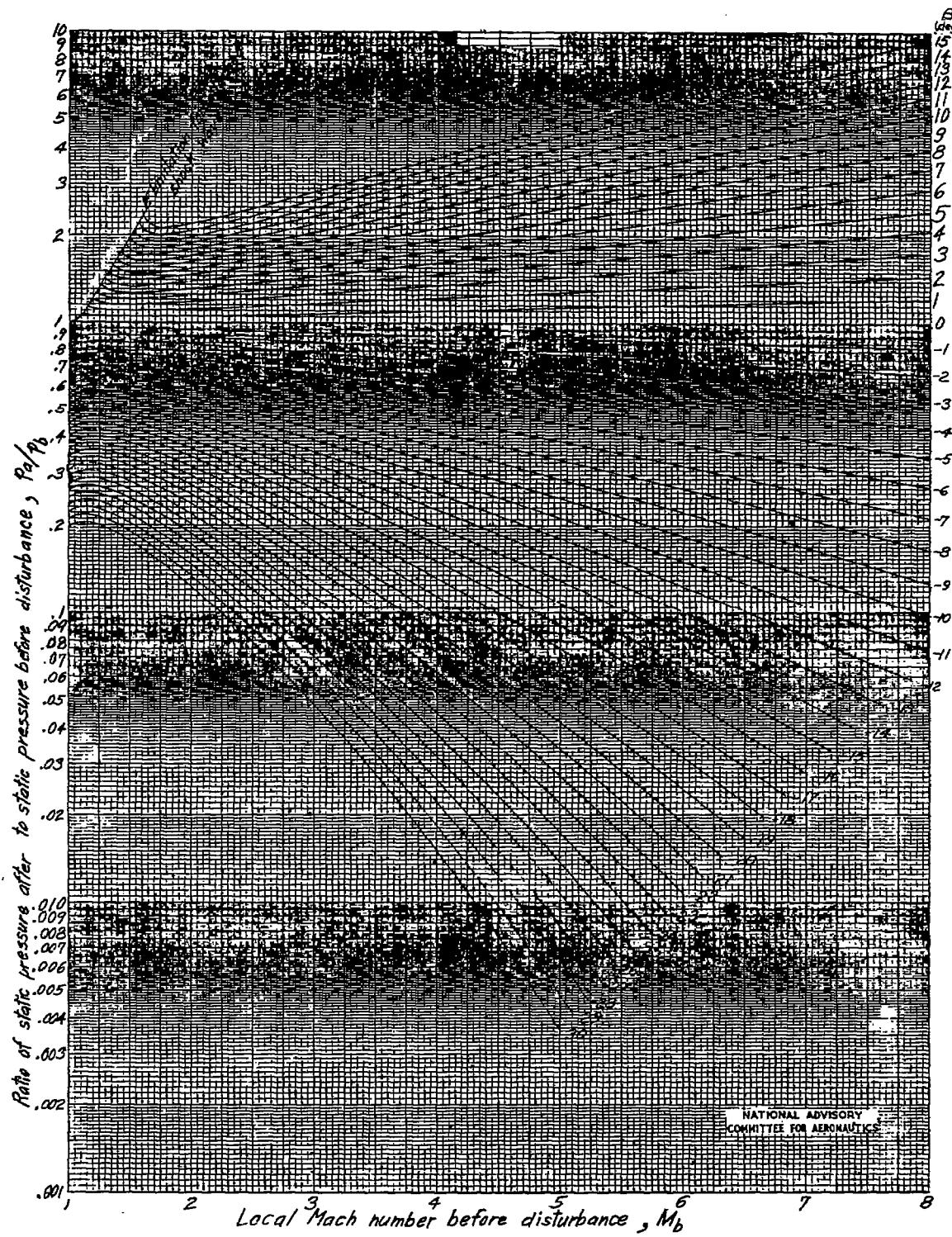


Figure 6. - Static pressure ratio across shock and expansion waves.

FIG. 7

NACA TN No. 1143

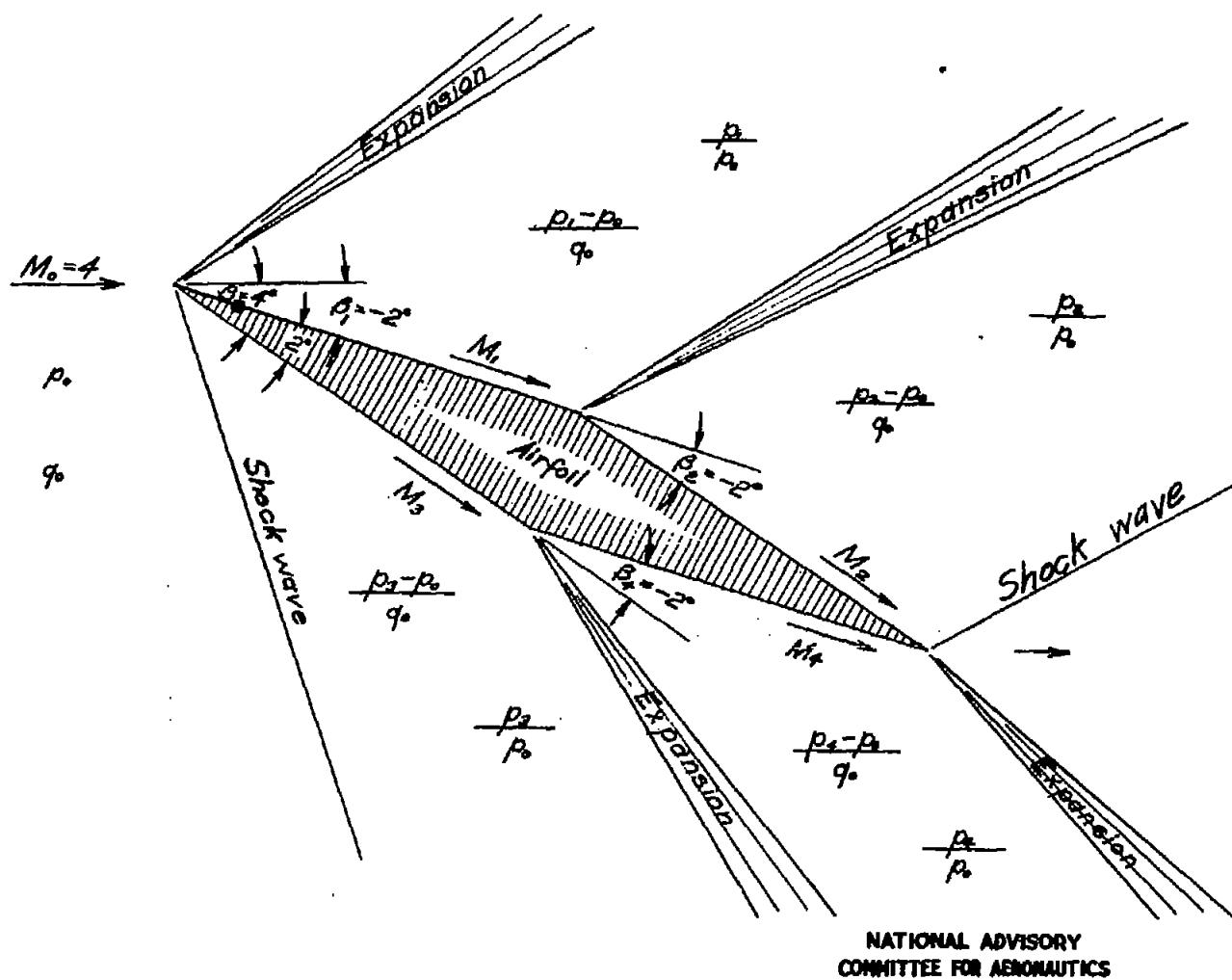


Figure 7.- Example airfoil (showing conditions to be determined).

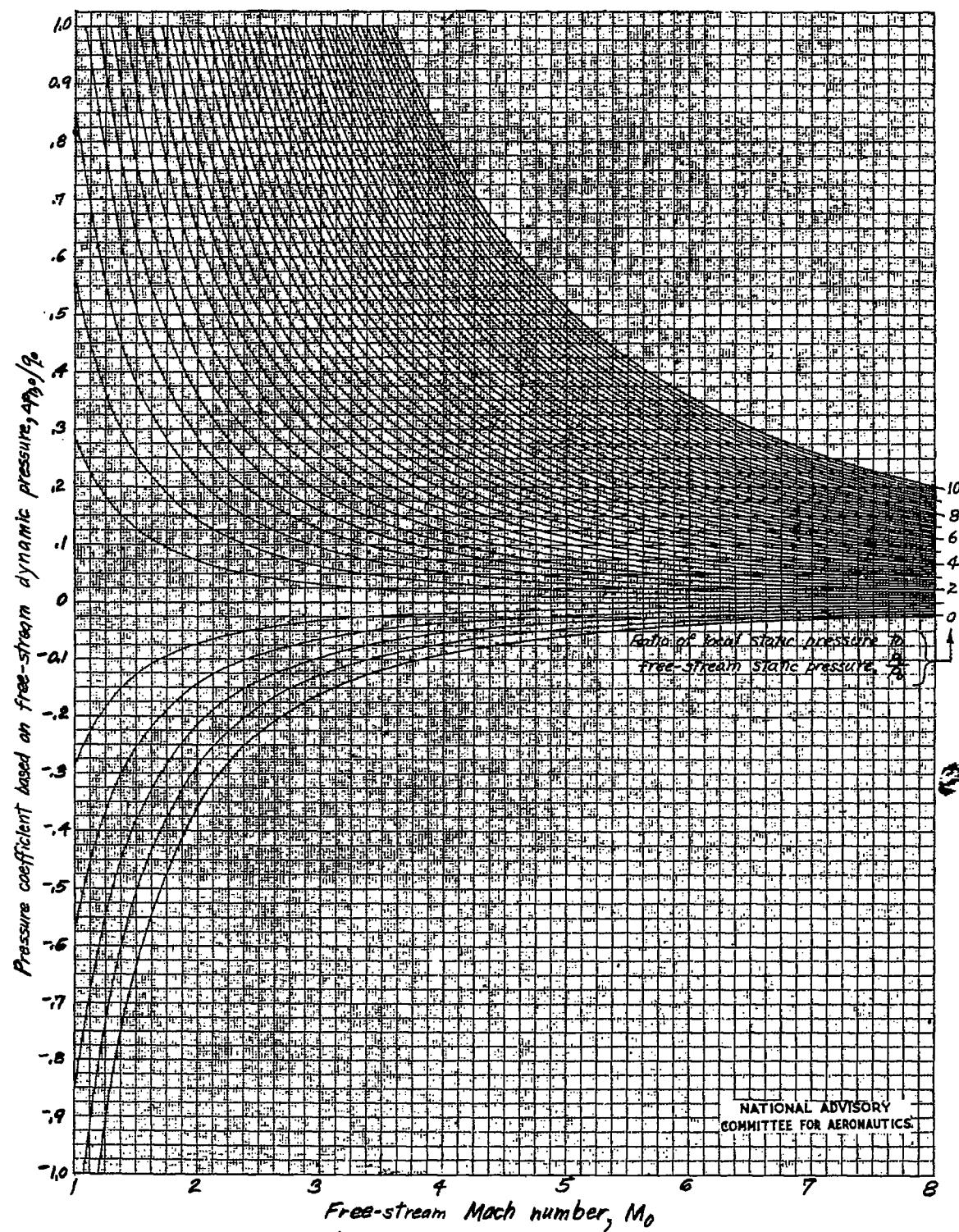


Fig. 9

NACA TN No. 1143

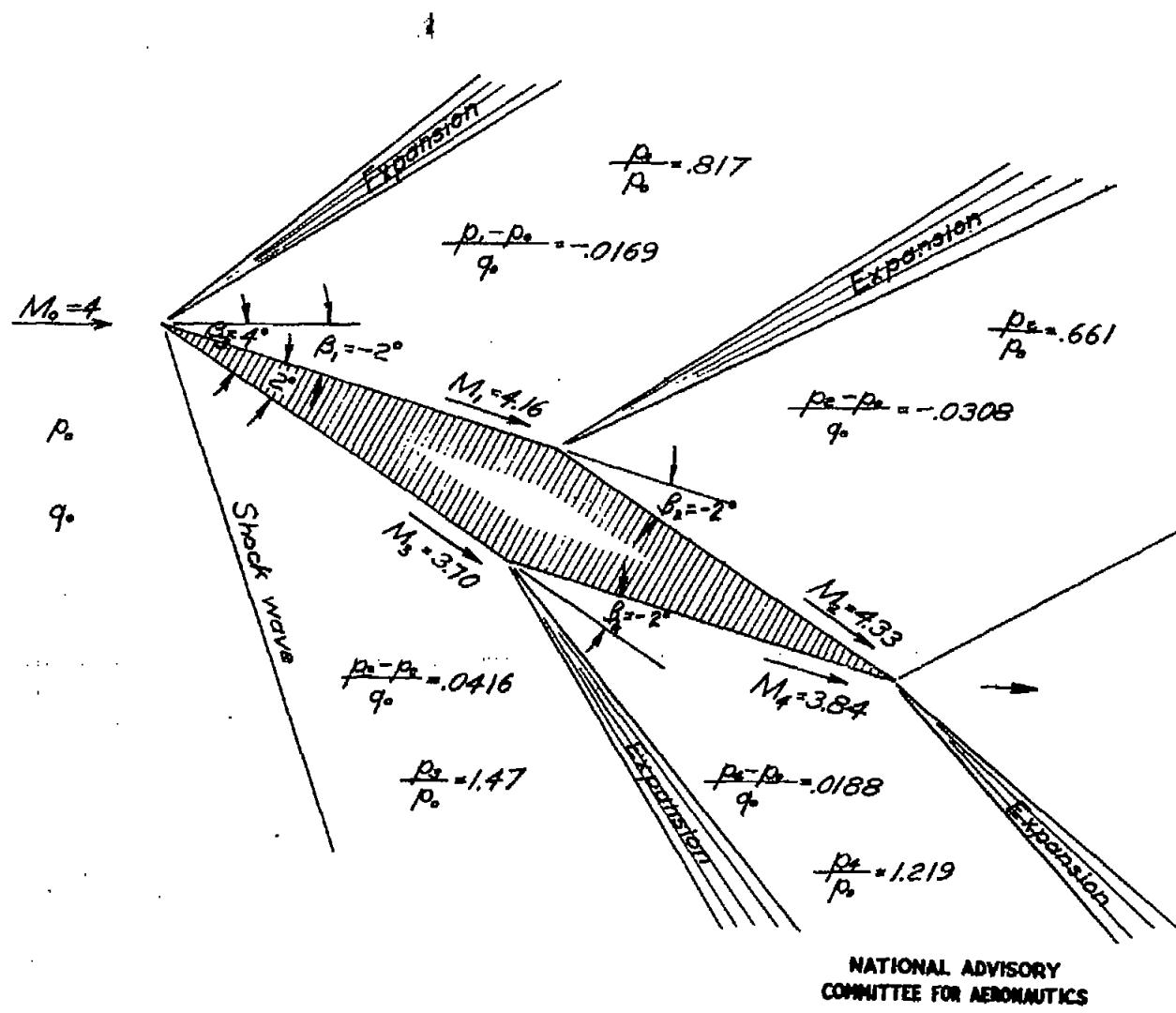


Figure 9.— Example airfoil (showing results obtained).

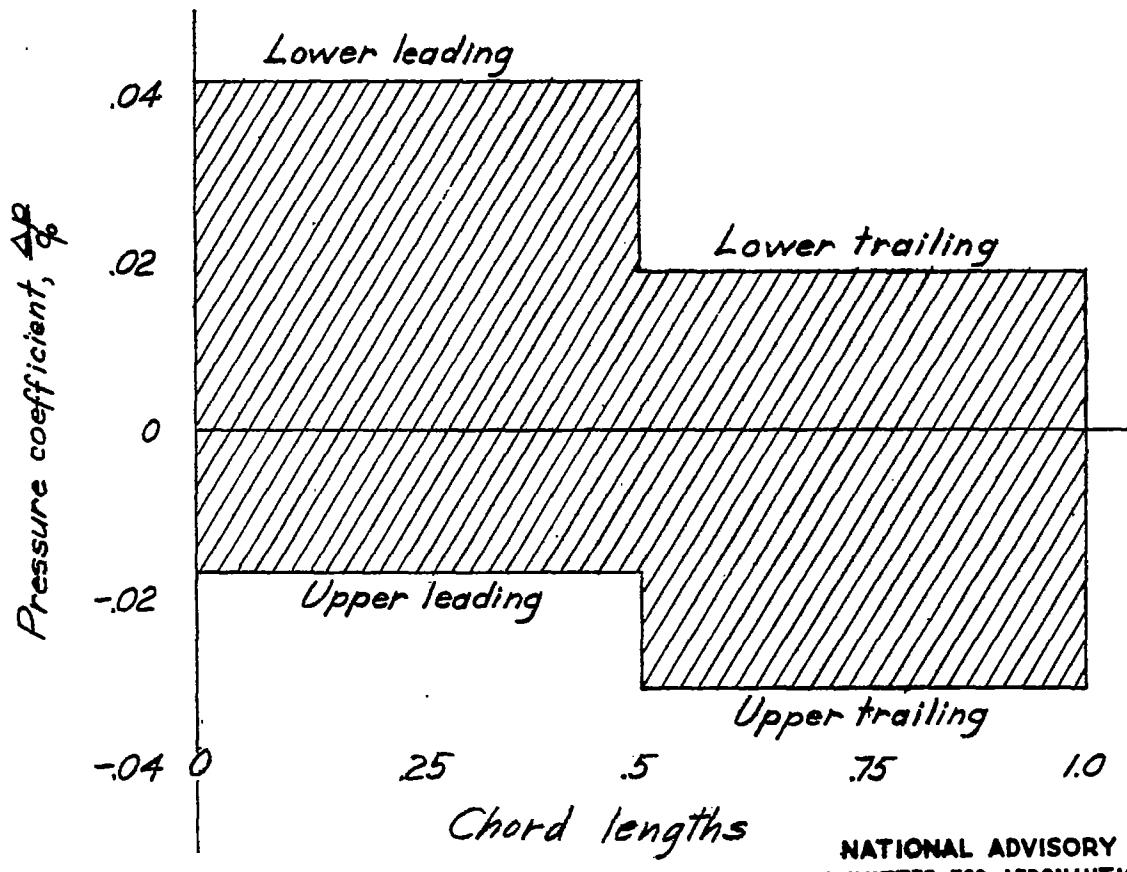


Figure 10.- Determination of lift coefficient from pressure distribution for example airfoil of figures 7 and 9. Value obtained by integrating shaded area gives lift coefficient, 0.0540.

FIG. 11

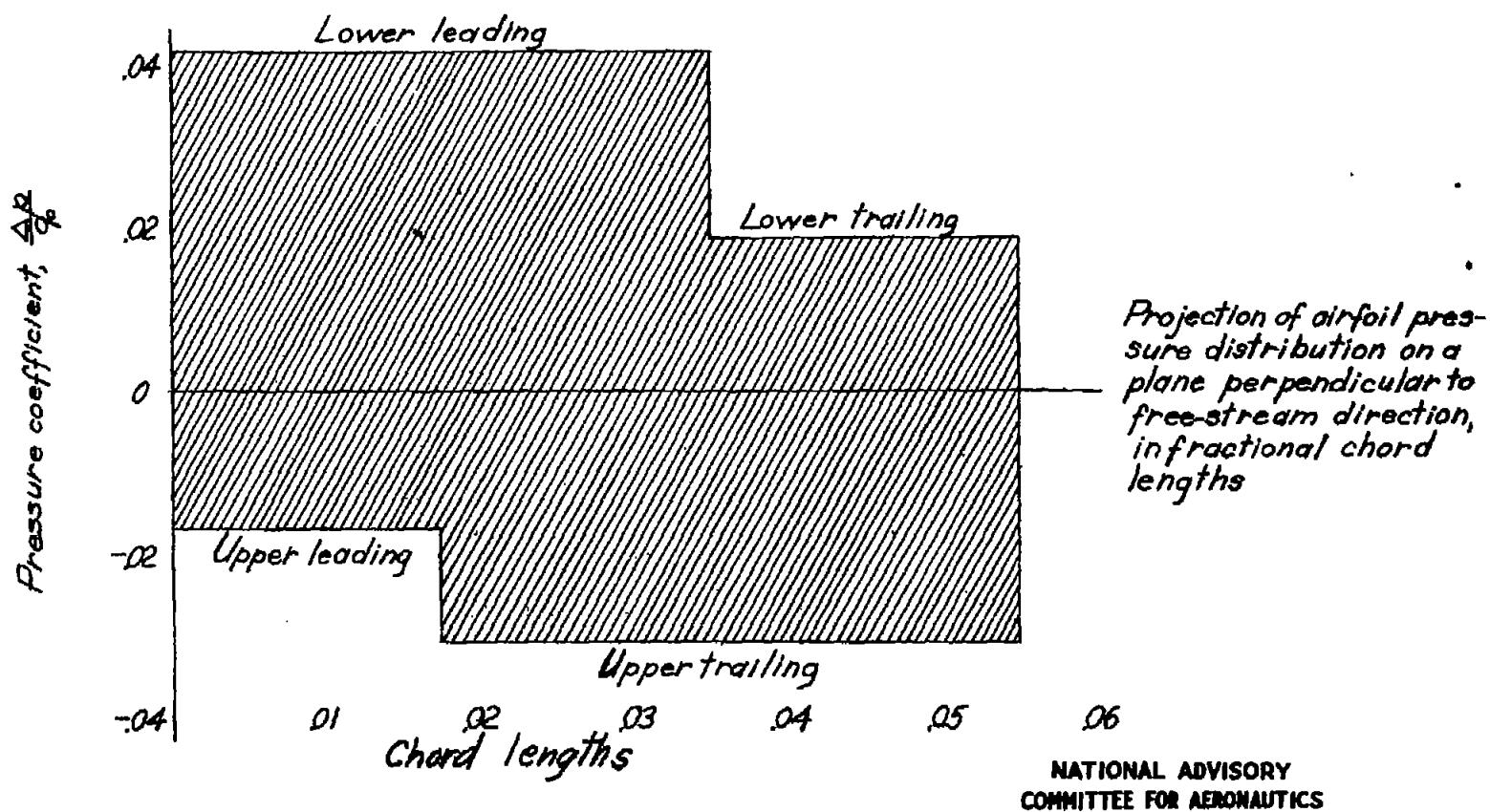


Figure 11.- Determination of drag coefficient from pressure distribution for example airfoil of figures 7 and 9. Value obtained by integrating shaded area gives pressure drag coefficient, 0.00315.

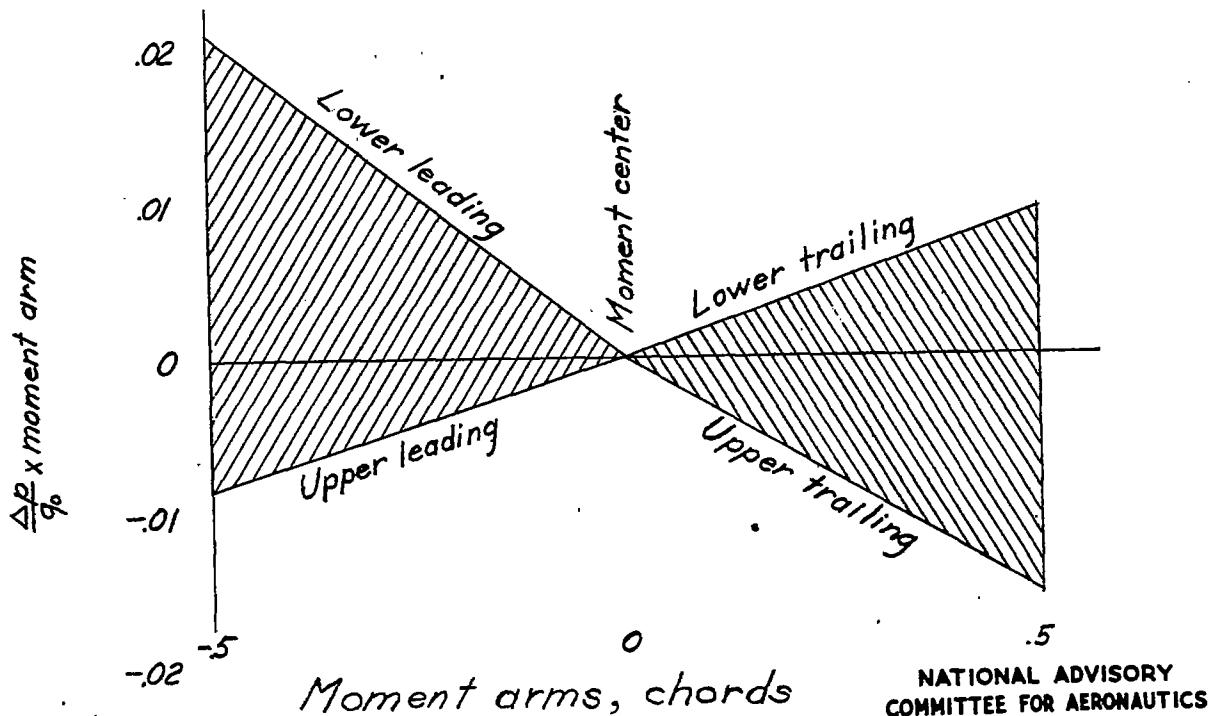


Figure 12.- Determination of moment coefficient from pressure distribution for example airfoil of figures 7 and 9. If leading and trailing surfaces give moments in the same sense add the area between "upper leading" and "lower leading" to that between "upper trailing" and "lower trailing" lines. The value obtained by integrating shaded area gives the moment coefficient about 0.50 chord, 0.0001112.

Fig. 13

NACA TN No. 1143

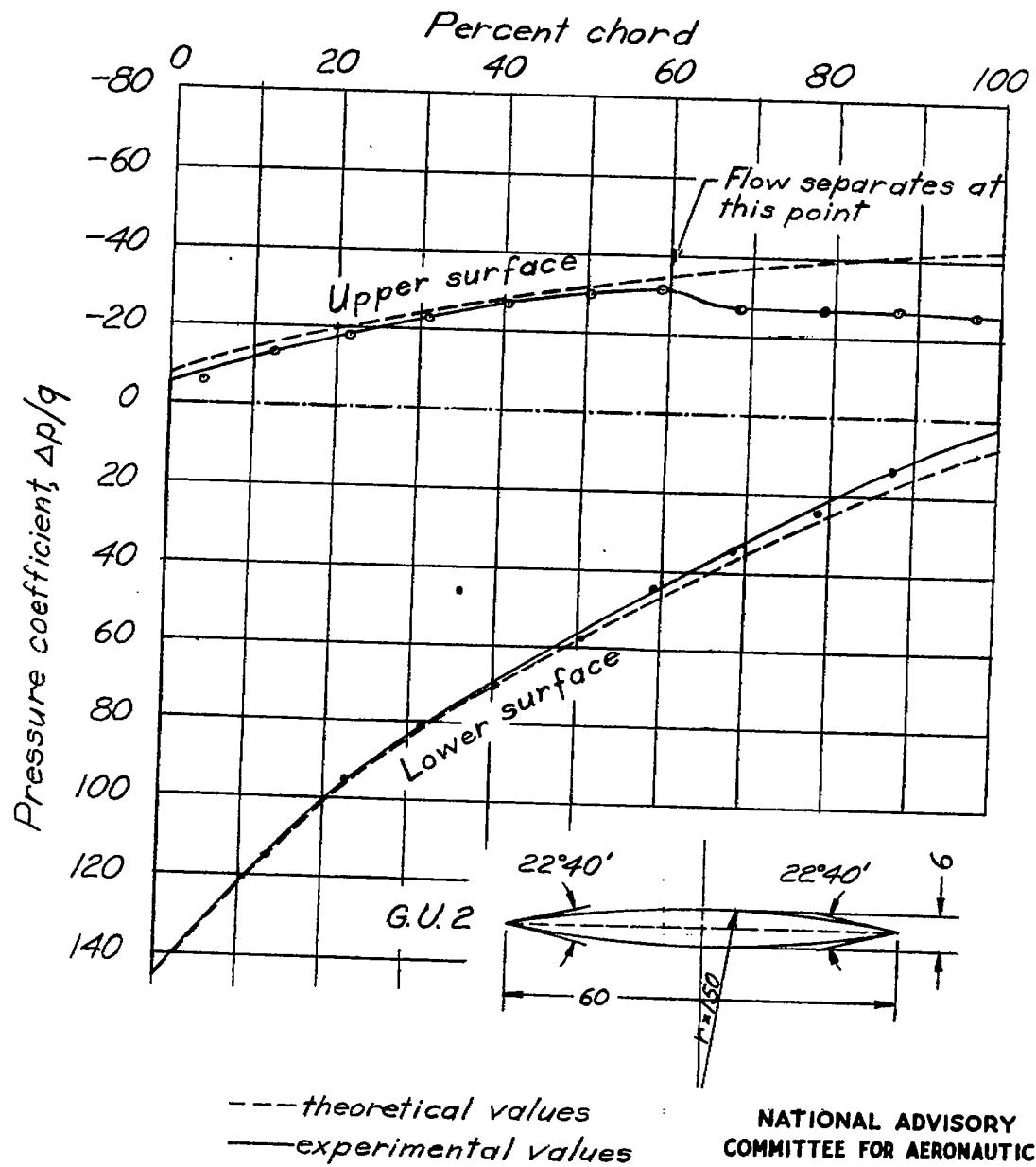


Figure 13.— Comparison of calculated and experimental results from NACA TM No. 946. $\alpha = 14^\circ$; $M_\infty = 2.13$; $R = 640,000$; thickness = 0.10 chord.

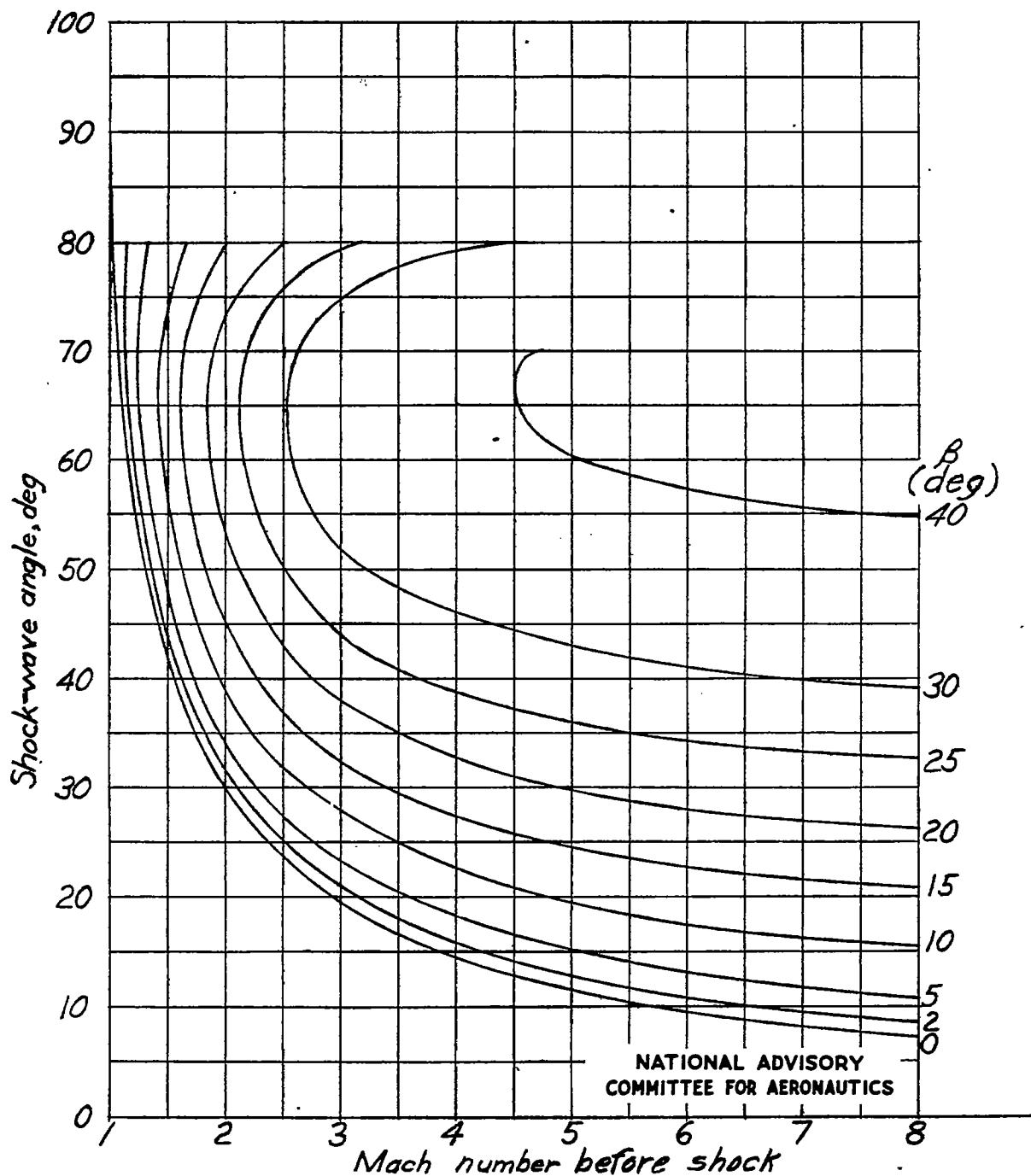


Figure 14.- Shock-wave angle for air.

NACA

SUPPLEMENT

NACA TECHNICAL NOTE NO. 1143

CHARTS FOR DETERMINING THE CHARACTERISTICS OF
SHARP-NOSE AIRFOILS IN TWO-DIMENSIONAL
FLOW AT SUPERSONIC SPEEDS

By H. Reese Ivey, George W. Stickle,
and Alberta Schuettler

September 1947

Since Technical Note No. 1143 was completed, the need for a more extensive version of table I, "Values of Local Mach Number, Pressure Ratio, and Pressure Coefficient across Shock Waves," has become apparent. This table is now available in expanded form; and a copy of the expanded table is included in this supplement to supersede the original table I.

Errors in the original publication are as follows:

Page 11: The first sentence of the last paragraph should begin "Tables I and III . . ." instead of "Tables I and II . . .".

Corrections in tables II and III are as follows:

Table II.--

M_b	$-\beta$	M_a
2.3	23°	3.4225
2.6	10°	3.0867
4.3	3°	4.5658
4.7	7°	5.4669
4.7	8°	5.5922
4.7	9°	5.7240
6.2	9°	7.9200

Table III.--

M_b	$-\beta$	p_a/p_b
1.4	14°	0.49071
4.7	24°	.02350

Figure 13.- Each vertical space along the scale label for pressure coefficient $\Delta p/q$ should represent 0.125 instead of 20; thus, the vertical scale should appear:

-.500
-.375
-.250
-.125
0
.125
.250
.375
.500
.625
.750
.875
1.000

TABLE I - VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND
PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Continued

θ (deg)	β (deg)	M_b	M_a	P_a P_b	$\frac{\rho_{a,b}}{\rho_b}$	θ (deg)	β (deg)	M_b	M_a	P_a P_b	$\frac{\rho_{a,b}}{\rho_b}$
36	27	6.85490	2.77749	18.77364	0.54035	39	17	2.51649	1.79783	2.75938	0.39689
	28	9.49197	2.93232	36.14927	0.5732		18	2.61643	1.82749	2.99641	.41661
37	0	1.66164	1.66164	1.00000	0		19	2.72771	1.88043	3.27119	.43607
1	1.66897	1.66897	1.05244	0.02997		20	2.85280	1.89684	3.29373	.45529	
2	1.73706	1.66883	1.10831	0.02182		21	2.99499	1.93714	3.37713	.47427	
3	1.77776	1.67443	1.16800	0.07980		22	3.15879	1.96165	4.44366	.49304	
4	1.81934	1.68140	1.23196	0.10011		23	3.35052	2.03163	5.02033	.51161	
5	1.86352	1.68980	1.30070	0.12370		24	3.68530	2.09759	6.03867	.52999	
6	1.91000	1.69967	1.37483	0.14678		25	3.89991	2.14968	6.71711	.54821	
7	1.95907	1.71107	1.44505	0.16938		26	4.21505	2.22019	8.04246	.56627	
8	2.01103	1.72408	1.51421	0.19153		27	4.68611	2.30037	9.87974	.58418	
9	2.06623	1.73879	1.63730	0.21329		28	5.15512	2.39221	13.08374	.60196	
10	2.12508	1.75530	1.74154	0.23458		29	6.18247	2.49893	16.86668	.61961	
11	2.18589	1.77737	1.85637	0.25952		30	8.52952	2.62389	33.44886	.63716	
12	2.25584	1.79422	1.98398	0.27612							
13	2.38903	1.81694	2.12538	0.29638							
14	2.40896	1.84209	2.28541	0.30843							
15	2.44541	1.86986	2.46456	0.33599							
16	2.50099	1.90094	2.66997	0.35937							
17	2.69649	1.93445	2.90670	0.37449							
18	2.81543	1.97195	3.18289	0.39337							
19	2.94931	2.01330	3.50881	0.42003							
20	3.10242	2.05963	3.90033	0.43048							
21	3.26008	2.11101	4.37946	0.44973							
22	3.48991	2.16844	4.97971	0.46679							
23	3.74331	2.23295	5.73420	0.48469							
24	4.05245	2.30533	6.79240	0.50284							
25	4.46512	2.38873	8.29773	0.58004							
26	5.02050	2.48343	10.48377	0.53751							
27	5.81650	2.59406	14.27360	0.59487							
28	7.27313	2.72345	22.17432	0.72121							
29	10.76973	2.87776	48.84307	0.59247							
38	0	1.64247	1.64247	1.00000	0						
1	1.66000	1.66161	1.05191	0.02691							
2	1.69720	1.68923	1.10711	0.05312							
3	1.73592	1.63363	1.16955	0.07867							
4	1.77646	1.63934	1.22888	0.10361							
5	1.81891	1.64639	1.29635	0.12796							
6	1.86348	1.68184	1.36982	0.15177							
7	1.91042	1.66174	1.44727	0.17507							
8	1.95999	1.67607	1.53211	0.19758							
9	2.01354	1.68903	1.62841	0.22023							
10	2.06839	1.70361	1.72324	0.24217							
11	2.12800	1.71998	1.83985	0.26359							
12	2.19194	1.73287	1.95797	0.28424							
13	2.26069	1.75849	2.09371	0.30562							
14	2.33512	1.77219	2.24461	0.32607							
15	2.41504	1.80577	2.41463	0.34621							
16	2.50164	1.83318	2.50744	0.36607							
17	2.60229	1.86344	2.68793	0.38662							
18	2.71076	1.89686	3.03860	0.40498							
19	2.83337	1.93381	3.38091	0.42958							
20	2.97018	1.97473	3.73452	0.44261							
21	3.12334	2.02015	4.16104	0.46413							
22	3.18564	2.07076	4.6894	0.47953							
23	3.53153	2.12783	5.34815	0.49809							
24	3.79744	2.19070	6.21029	0.51625							
25	4.12302	2.26038	7.37982	0.53107							
26	4.56731	2.34390	9.02403	0.55184							
27	5.17392	2.43743	11.67110	0.56947							
28	6.10326	2.56545	16.30565	0.58669							
29	7.80868	2.67596	26.79741	0.60440							
30	12.89714	2.82462	73.38923	0.62171							
39	0	1.59902	1.58902	1.00000	0						
1	1.62367	1.58983	1.05145	0.02788							
2	1.66967	1.59190	1.10603	0.05500							
3	1.69713	1.59517	1.14616	0.08142							
4	1.73622	1.59970	1.22627	0.10718							
5	1.77709	1.60530	1.29551	0.13832							
6	1.81992	1.61261	1.36371	0.15687							
7	1.84695	1.62107	1.44036	0.18087							
8	1.91839	1.63094	1.52117	0.20436							
9	1.96255	1.64227	1.61896	0.22135							
10	2.01575	1.65214	1.71076	0.24989							
11	2.07236	1.66963	1.82770	0.27400							
12	2.13857	1.68984	1.93232	0.29370							
13	2.19774	1.70390	2.06908	0.31501							
14	2.25769	1.72394	2.20939	0.33597							
15	2.34343	1.74612	2.37081	0.35699							
16	2.42599	1.77061	2.55271	0.37689							

TABLE I.—VALUES OF LOCAL Mach NUMBER, PRESSURE RATIO, AND
PRESSURE COEFFICIENT ACROSS SHOCK WAVES — Continued

θ (deg)	β (deg)	M_0	M_∞	P_A P_D	$\frac{\Delta P_{A,b}}{P_D}$	θ (deg)	β (deg)	M_0	M_∞	P_A P_D	$\frac{\Delta P_{A,b}}{P_D}$
72	35	3.43290	0.76637	12.26933	1.36609	74	26	2.61118	0.70929	7.29665	1.29930
	36	3.61351	.76988	13.84205	1.38197		29	2.74053	.70845	7.52987	1.31813
	37	3.89188	.77395	15.81213	1.39765						
	38	4.18935	.77859	18.35382	1.41255						
	39	4.55839	.78383	21.76092	1.42730		30	2.86021	.70805	8.69246	1.33631
	40	5.03335	.78869	26.56793	1.44173						
	41	5.67920	.79221	31.86900	1.45504						
	42	6.63532	.80542	46.29378	1.46966						
	43	8.27973	.81131	72.17662	1.48320						
73	0	1.04569	1.04569	1.00000	0	75	35	3.70963	.71292	14.66843	1.41893
	1	1.05484	1.01397	1.08699	1.08002		36	3.95507	.71589	16.78182	1.43403
	2	1.12402	0.95560	1.18132	1.20508		37	4.27222	.71815	19.50336	1.44873
	3	1.16133	.96009	1.27725	1.29267		38	4.65280	.72151	23.16702	1.46304
	4	1.20286	.97075	1.37706	1.37299		39	5.11159	.72539	28.53207	1.47699
	5	1.24271	.91618	1.48103	1.44499		40	5.80587	.72980	36.18910	1.49061
	6	1.29898	.89721	1.58095	.51166		41	6.79229	.73477	49.56336	1.50391
	7	1.32373	.87993	1.70293	.57307		42	8.49002	.74033	77.53807	1.51692
	8	1.36112	.86116	1.82122	.62904						
	9	1.40719	.84974	1.94606	.68222						
	10	1.45006	.83908	2.07671	.73156		0	1.03526	1.03728	1.00000	0
	11	1.49385	.82448	2.21341	.77735		1	1.07856	.99910	1.09961	1.12232
	12	1.53553	.81330	2.35984	.80009		2	1.12179	.96707	1.20313	1.23060
	13	1.58466	.80333	2.51265	.86019		3	1.16908	.93853	1.31088	1.37118
	14	1.63193	.79409	2.67180	.89838		4	1.20854	.91844	1.42319	1.41392
	15	1.68065	.78566	2.84700	.93414		5	1.25231	.88990	1.54043	.99229
	16	1.73098	.77759	3.03020	.96796		6	1.29649	.86906	1.66300	.96118
	17	1.78310	.77102	3.22962	1.00000		7	1.34120	.85016	1.79136	.98118
	18	1.83722	.76473	3.43466	1.03043		8	1.38653	.83896	1.92601	.98110
	19	1.89371	.75907	3.65994	1.05937		9	1.43266	.81727	2.06732	1.74301
	20	1.95239	.75401	3.90033	1.08696						
	21	2.01600	.74933	4.15104	1.11330						
	22	2.07872	.74560	4.44566	1.13859						
	23	2.14693	.74220	4.73297	1.16023						
	24	2.21314	.73932	5.08756	1.18376						
	25	2.28982	.73693	5.15696	1.20799						
	26	2.35763	.73532	5.26838	1.22933						
	27	2.42690	.73466	5.3798	1.24958						
	28	2.50560	.73274	5.62480	1.26562						
	29	2.68217	.73229	7.39191	1.28903						
	30	2.77382	.73231	8.04246	1.30758						
	31	2.86647	.73260	8.78447	1.32554						
	32	3.03331	.73376	9.64373	1.34284						
	33	3.18421	.73520	10.65120	1.35562						
	34	3.35594	.73713	11.84954	1.37621						
	35	3.55267	.73954	13.29968	1.39215						
	36	3.78165	.74246	15.09256	1.40766						
	37	4.05344	.74950	17.36355	1.42276						
	38	4.28115	.74988	20.34079	1.43749						
	39	4.49990	.75441	24.41461	1.45206						
	40	5.34841	.75951	30.33228	1.46389						
	41	6.11240	.76223	39.1928	1.47962						
	42	7.13180	.77158	56.90169	1.49304						
	43	9.81869	.77860	98.56608	1.50619						
	44	1.20194	.78214	1.38951	.39211						
	45	1.24461	.80321	1.50868	.46756						
	46	1.28668	.88333	1.62326	.53656						
	47	1.33227	.86925	1.74300	.59064						
	48	1.37446	.84878	1.86994	.65783						
	49	1.41584	.83373	2.00221	.71163						
	50	1.46318	.81997	2.14129	.76155		0	1.03061	1.03061	1.00000	0
	51	1.50891	.80738	2.28780	.80602		1	1.07648	.99179	1.10613	.13086
	52	1.55773	.79566	2.44246	.85141		2	1.12261	.97764	1.21660	.23571
	53	1.60358	.78518	2.60914	.89190		3	1.16798	.92137	1.33171	.34738
	54	1.65347	.77568	2.77996	.93021		4	1.21390	.90036	1.45105	.43606
	55	1.70411	.76684	2.96390	.96611		5	1.26011	.87613	1.57743	.51950
	56	1.75676	.75956	3.16033	1.00000		6	1.30675	.85928	1.69333	.59309
	57	1.81132	.75146	3.37020	1.03024		7	1.35393	.83451	1.84682	.65993
	58	1.86801	.74581	3.59205	1.06241		8	1.40180	.81657	1.99171	.72096
	59	1.92708	.73881	3.83673	1.09125		9	1.45048	.80023	2.11422	.77694
	60	1.98583	.73441	4.09738	1.11862		10	1.50012	.78532	2.30509	.88850
	61	2.05355	.72856	4.37946	1.14422		11	1.55087	.77170	2.47511	.87616
	62	2.12164	.72431	4.62944	1.16978		12	1.60284	.75924	2.65520	.92039
	63	2.19353	.72056	5.02033	1.18565		13	1.66266	.74783	2.81642	.96156
	64	2.26971	.71733	5.36868	1.21551		14	1.71128	.73738	3.04994	1.00000
	65	2.35078	.71462	5.70972	1.23545		15	1.76783	.72770	3.26602	1.03583
	66	2.43745	.71231	6.23611	1.25511		16	1.82665	.71906	3.49958	1.06900
	67	2.53058	.71060	6.73694	1.27978		17	1.88212	.71106	3.71609	1.10162
	68	2.62717					18	1.93183	.70276	4.01779	1.13164
	69						19	2.01842	.69711	4.30816	1.16002
	70						20	2.08825	.69209	4.60311	1.18693
	71						21	2.15171	.68565	4.96621	1.21247
	72						22	2.23938	.68076	5.34199	1.23676

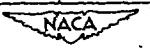


TABLE I.—VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND
PRESSURE COEFFICIENT ACROSS SHOCK WAVES—Continued

θ (deg)	β (deg)	M_b	M_a	P_a P_b	$\frac{\Delta P_{a,b}}{P_b}$	θ (deg)	β (deg)	M_b	M_a	P_a P_b	$\frac{\Delta P_{a,b}}{P_b}$
80	26	3.36182	.56747	12.68123	1.44934	83	0	1.00751	1.00751	1.00000	0
	27	3.57242	.58403	14.27360	1.44936		1	1.09383	.93218	1.20849	.24893
	28	3.61551	.58699	15.30565	1.50193		2	1.17888	.87206	1.43090	.44866
	29	4.10144	.59833	18.88699	1.51733		3	1.26372	.82273	1.66983	.59879
	30	4.44589	.58610	22.19784	1.53209		4	1.34874	.76138	1.92411	.72572
	31	4.87369	.59421	26.70936	1.54624		5	1.43464	.74614	2.19890	.83214
	32	5.42791	.59269	33.15942	1.55884		6	1.52198	.71570	2.49570	.92342
	33	6.19057	.59153	43.19544	1.57292		7	1.61348	.68911	2.81748	1.00000
	34	7.34537	.59072	60.88809	1.58932		8	1.70328	.66568	3.18775	1.06743
	35	9.43104	.59025	100.474298	1.59767		9	1.79843	.64487	3.55072	1.12661
	36	1.02127	1.02127	1.00000	0		10	1.89748	.62627	3.97144	1.17901
	37	1.08039	.99273	1.16228	.38933		11	2.00118	.60935	4.13611	1.22274
	38	1.14803	.90300	1.33333	.36130		12	2.11041	.59447	4.95229	1.26770
	39	1.21520	.86083	1.51400	.49725		13	2.22621	.56880	5.52947	1.30562
	40	1.28248	.88497	1.70285	.62266		14	2.34981	.56838	6.17952	1.34007
81	3	1.35020	.79307	1.90817	.73166		15	2.48270	.52703	6.91764	1.37152
	4	1.41868	.76527	2.12398	.79773		16	2.6277	.49671	7.16366	1.40036
	5	1.48524	.74071	2.35410	.87339		17	2.78434	.47472	8.14365	1.42623
	6	1.55918	.71880	2.60016	.96031		18	2.95847	.44598	9.59398	1.45149
	7	1.63184	.69915	2.86404	1.00000		19	3.15177	.42604	11.26604	1.47438
	8	1.70656	.68144	3.14793	1.05361		20	3.37388	.31336	12.91636	1.49550
	9	1.78371	.66942	3.42440	1.10204		21	3.62824	.30688	14.96338	1.51531
	10	1.86570	.65087	3.76846	1.14604		22	3.92270	.30018	17.51892	1.53360
	11	1.94699	.63763	4.14761	1.18621		23	4.28813	.29496	20.96744	1.55127
	12	2.03408	.62954	4.54226	1.22306		24	4.73021	.29261	23.63151	1.56767
	13	2.12218	.61449	4.97544	1.26969		25	5.38518	.28517	32.42973	1.58314
	14	2.21059	.60438	5.43345	1.28837		26	6.14449	.28093	43.22632	1.59777
	15	2.30469	.59511	5.98393	1.31758		27	7.14895	.27703	63.09215	1.61163
	16	2.40400	.58662	6.57644	1.34484		28	9.86743	.27321	111.74018	1.62880
	17	2.50155	.57883	7.24294	1.36988		29	1.00551	1.00000	0	0
	18	2.60778	.57161	7.99023	1.39324		30	1.10495	.91901	1.24370	.28484
	19	2.61800	.56517	8.68534	1.41582		31	1.20403	.89207	1.50615	.49878
	20	2.69687	.56229	9.86553	1.43877		32	1.30210	.79335	1.78977	.66545
	21	3.13708	.55376	11.03386	1.45623		33	1.40075	.75410	2.09743	.79902
	22	3.32993	.54882	12.42288	1.47501		34	1.50063	.71689	2.43052	.90573
82	3	3.54049	.54434	14.09971	1.49892		35	1.60119	.68510	2.49515	.90000
	4	3.78811	.54030	15.16503	1.50973		36	1.70625	.67758	3.26220	1.07728
	5	4.07944	.53669	18.77364	1.55973		37	1.80513	.63350	3.44711	1.14426
	6	4.43060	.53346	22.17482	1.58498		38	1.90528	.61223	4.14307	1.20221
	7	4.88743	.53063	26.79740	1.59553		39	2.03312	.59336	4.69748	1.25307
	8	5.43471	.52817	33.44887	1.56915		40	2.18107	.57646	5.32259	1.28610
	9	6.21880	.52607	43.84824	1.58278		41	2.31797	.56185	6.03330	1.33886
	10	7.43589	.52433	62.49239	1.59557		42	2.46747	.54752	6.84903	1.37433
	11	9.61661	.52292	105.05334	1.60736		43	2.62651	.53507	7.79553	1.40692
	12	1.00983	1.00983	1.00000	0		44	2.80428	.52373	8.90771	1.43652
	13	1.05893	.91315	1.18238	.22096		45	3.00223	.51342	10.23417	1.46353
	14	1.16106	.88669	1.37561	.30804		46	3.22630	.50399	11.84448	1.48634
	15	1.23589	.84318	1.53082	.38223		47	3.48227	.49335	13.84605	1.51117
	16	1.31088	.80449	1.79930	.46149		48	3.78761	.48741	15.38741	1.53228
83	17	1.38645	.77094	2.03282	.76734		49	4.15390	.48019	19.74110	1.55187
	18	1.46304	.74807	2.26218	.80574		50	4.61161	.47341	24.37689	1.57010
	19	1.54104	.71648	2.529026	.93827		51	5.22824	.46744	31.18628	1.58712
	20	1.62087	.69379	2.83905	1.00000		52	6.04172	.46177	43.95108	1.60680
	21	1.70297	.67352	3.15185	1.05969		53	7.40162	.45673	63.04960	1.61803
	22	1.78780	.65533	3.49004	1.11294		54	10.10199	.45480	127.59103	1.63212
	23	1.87587	.63893	3.85920	1.16075		55	0	1.00382	1.00000	0
	24	1.96776	.62407	4.26328	1.20395		56	1.12309	.90238	1.89372	.33266
	25	2.06412	.61058	4.70714	1.24230		57	1.24028	.82702	1.61338	.37056
	26	2.16969	.59829	5.19927	1.27304		58	1.35784	.76288	1.96611	.47924
	27	2.27337	.58706	5.74613	1.31190		59	1.47518	.70931	2.35391	.58844
	28	2.38822	.57680	6.33862	1.34217		60	2.59637	.68174	2.78387	1.00000
	29	2.51150	.56740	7.01972	1.37016		61	3.72126	.64870	3.46362	1.09147
	30	2.64482	.55479	7.83617	1.39612		62	1.85137	.60041	3.80263	1.16765
	31	2.79013	.54083	8.73972	1.42029		63	1.98885	.55986	4.41303	1.23265
	32	2.94999	.53468	9.78928	1.44226		64	2.13493	.57436	5.11092	1.28834
	33	3.12533	.52694	11.00284	1.46373		65	2.29200	.55533	5.91560	1.33674
	34	3.32734	.52055	12.49953	1.48384		66	2.46216	.53283	6.84960	1.37922
	35	3.55522	.50299	13.29389	1.50232		67	2.69087	.52388	7.96937	1.41683
	36	3.81969	.50230	15.52903	1.52010		68	2.82089	.50964	9.30959	1.49037
	37	4.13307	.51573	19.37661	1.53682		69	3.09929	.49731	10.95474	1.48050
	38	4.51162	.51158	23.15148	1.55651		70	3.37530	.48612	13.02379	1.50771
	39	4.99598	.50783	28.38911	1.56761		71	3.70688	.47594	15.70664	1.53244
	40	5.63407	.50447	36.18497	1.58166		72	4.10319	.46666	19.38653	1.55501
	41	6.54508	.50149	43.84302	1.59934		73	4.61339	.45817	24.48172	1.57572
	42	8.01831	.49887	73.38928	1.60646		74	5.30513	.45040	32.41900	1.59479



TABLE I.— VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND
PRESSURE COEFFICIENT ACROSS SHOCK WAVES — Concluded

θ (deg)	β (deg)	M_b	M_a	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$	θ (deg)	β (deg)	M_b	M_a	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$
85	20	6.33047	0.44327	46.23264	1.61232	87	5	2.01891	0.58196	4.54790	1.88076
	21	8.12762	1.43674	76.31560	1.62078		6	2.02574	0.56055	5.73327	1.33165
86	0	1.00244	1.00244	1.00000	0	88	7	2.02093	0.51836	7.88629	1.40271
	1	1.15055	1.87994	1.37080	.39921		8	2.01902	1.03410	9.27715	1.45677
	2	1.25531	1.79401	1.78307	.66612		9	2.02349	1.47324	12.00248	1.50273
	3	1.44179	1.78927	2.84673	.05680		10	1.70118	.45907	15.94670	1.51176
	4	1.59062	1.67901	2.77455	1.00000		11	1.38111	.13210	22.18216	1.57589
	5	1.74555	1.63805	3.37082	1.22156		12	1.37519	.16193	33.44009	1.60442
	6	1.90850	1.60405	1.06208	1.20098		13	1.20410	.11232	50.21656	1.63599
	7	2.08259	.57329	1.86877	1.87488		14	1.00959	.80499	80.45147	1.67262
	8	2.27139	.39060	5.88313	1.33551		15	13.26355			
	9	2.47941	.38914	6.97050	1.39744		16				
87	10	2.71274	.51031	8.37701	1.43208	89	0	1.00061	1.00061	1.00000	0
	11	2.97592	.49365	10.14682	1.47087		1	1.29222	.79143	2.77959	.66653
	12	3.29358	.47880	18.42734	1.50491		2	1.28355	.67940	2.75534	1.00000
	13	3.67355	.46948	15.50082	1.53502		3	1.29445	.59921	4.01552	1.20085
	14	4.15329	.45349	19.86017	1.56194		4	2.84473	.54457	5.70487	1.33368
	15	4.79538	.44264	26.53113	1.38608		5	2.66202	.50254	8.09068	1.42944
	16	5.73559	.43879	38.08269	1.60789		6	3.19358	.46933	11.71754	1.50182
	17	7.35053	.42388	62.56602	1.62772		7	3.93752	.44219	17.89940	1.52718
	18	11.43333	.41364	151.51514	1.64988		8	2.15650	.41950	30.83550	1.60196
	19	1.79219	.62346	3.57035	1.14321		9	8.01993	.40080	74.70644	1.63670

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